

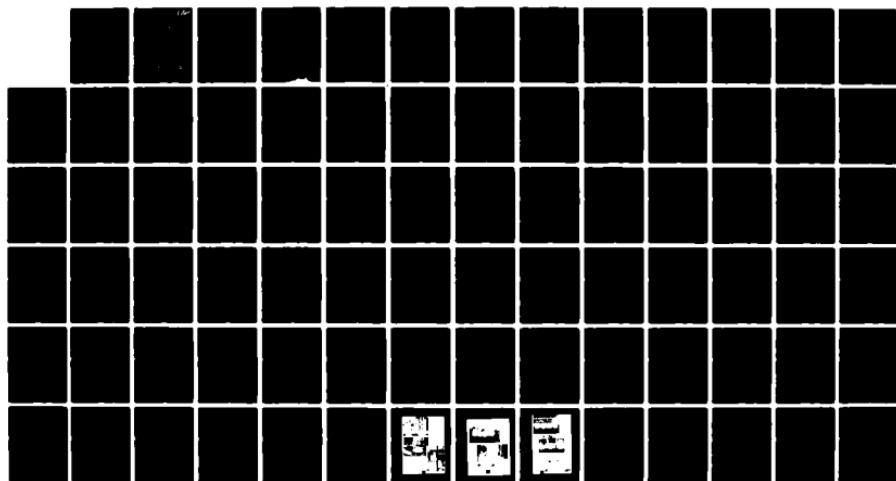
AD-A134 481 APPLICATION OF THERMOCHEMICAL MODELING TO AIRCRAFT  
INTERIOR POLYMERIC MAT. (U) JET PROPULSION LAB PASADENA  
CA W DOKKO ET AL. OCT 83 JPL-D-955 DDT/FAA/CT-83/16

1/1

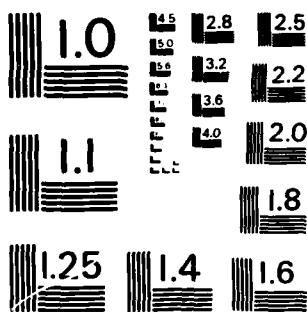
UNCLASSIFIED

F/G 1/2

NL



END  
DATE FILMED  
71-83  
DTIC



MICROCOPY RESOLUTION TEST CHART  
NATIONAL BUREAU OF STANDARDS-1963-A

12

DOT/FAA/CT-83/16

AD-A134 481

# **Application of Thermochemical Modeling to Aircraft Interior Polymeric Materials II - Multilayered Seat**

W. Dokko  
K. Ramohalli

Prepared by  
Jet Propulsion Laboratory  
California Institute of Technology  
Pasadena, California

October 1983

Final Report

This document is available to the U.S. public  
through the National Technical Information  
Service, Springfield, Virginia 22161.

DTIC FILE COPY



US Department of Transportation  
Federal Aviation Administration  
Technical Center  
Atlantic City Airport, N.J. 08405

DTIC  
ELECTED  
NOV 8 1983  
S D  
D

83 - 11 07 015

**NOTICE**

This document is disseminated under the sponsorship of the Department of Transportation in the interest of information exchange. The United States Government assumes no liability for the contents or use thereof.

The United States Government does not endorse products or manufacturers. Trade or manufacturer's names appear herein solely because they are considered essential to the object of this report.

## Technical Report Documentation Page

1. Report No. DOT/FAA/CT-83/16	2. Government Accession No. <i>AD-A134 481</i>	3. Recipient's Catalog No.	
4. Title and Subtitle <b>Application of Thermochemical Modeling to Aircraft Interior Polymeric Materials, II - Multilayered Seat Cushions</b>		5. Report Date October 1983	
7. Author(s) Won Dokko and Kumar Ramohalli		6. Performing Organization Code	
9. Performing Organization Name and Address Jet Propulsion Laboratory California Institute of Technology Pasadena, California 91109		8. Performing Organization Report No. JPL D-955	
12. Sponsoring Agency Name and Address U.S. Department of Transportation Federal Aviation Administration Technical Center Atlantic City Airport, New Jersey 08405		10. Work Unit No. (TRAIL) 11. Contract or Grant No. 181-350-100	
15. Supplementary Notes		13. Type of Report and Period Covered June 1981 - May 1982	
16. Abstract This report summarizes the results from a twelve-month study of applying thermochemical modeling to multilayered polymeric materials, passenger aircraft seat cushions. The use of fire-blocking layer(s) between the foam cushion and the covering fabric has been studied extensively to minimize fire hazards from aircraft seats. The objectives of this work are to expand the thermochemical model for the multilayered materials and to experimentally verify theoretical predictions. First, the thermochemical model is extended to any number of multilayered materials, by applying the same analysis technique used in the previous work. The additional constraints of temperature and heat flux continuities at every interface are also applied. A computer program is developed to predict burning behavior of seat cushion systems with and without a fire-blocking layer. Second, a series of tests burning seat cushions with and without a fire-blocking layer are conducted in a modified NBS Smoke Density Chamber. The chamber was supplemented with a multichannel recorder/multiple thermocouples and a weight-measuring device to determine temperature profile and to continuously monitor weight loss, respectively. The results indicate that the predicted temperature profiles are in very good agreement with the experimentally determined ones, and that the same effectiveness of the fire-blocking layers are predicted as those of actual weight loss measurements. It is, however, observed that the formation and presence of a void inside of the polyurethane foam seem to cause the over-prediction of the temperature profile and under-prediction of the weight loss (compared to the case when the void is small or nonexistent).			
17. Key Words Thermochemical modeling; fire safety; aircraft interior materials, multilayers, fire blocking layer, seat cushion, temperature profile, weight loss	18. Distribution Statement This document is available to the U.S. public through the National Technical Information Service, Springfield, Virginia 22161		
19. Security Classif. (of this report) Unclassified	20. Security Classif. (of this page) Unclassified	21. No. of Pages	22. Price

NOTICE

This document is disseminated under the sponsorship of the Federal Aviation Administration, U. S. Department of Transportation in the interest of information exchange. The United States Government assumes no liability for the contents or use thereof.

The United States Government does not endorse products or manufacturers. Trade or manufacturers' names appear herein solely because they are considered essential to the object of this report.

Accession For	
NTIS GRA&I	<input checked="" type="checkbox"/>
DTIC TAB	<input type="checkbox"/>
Unannounced	<input type="checkbox"/>
Justification	
By _____	
Distribution/ _____	
Availability Codes	
Avail and/or	
Dist	Special
R/1	



PREFACE

The work described in this report was performed by the Jet Propulsion Laboratory (JPL), California Institute of Technology (Caltech), and was sponsored by the Federal Aviation Administration (FAA) through an agreement with the National Aeronautics and Space Administration (NASA). The authors would like to thank Dr. Thor Eklund, the FAA Project Manager, Messrs. Constantine Sarkos, Wayne Howell, and Richard Kirsch of the FAA and Allen Tobiason of NASA for their suggestions and support.

## TABLE OF CONTENTS

	<u>PAGE</u>
1. INTRODUCTION . . . . .	1
2. THEORETICAL . . . . .	2
(a) Applicability of the Steady-State Model . . . . .	2
(b) Mathematical Formulation . . . . .	3
(c) Computer Program . . . . .	10
(d) The Overall Logic and Characteristics of the Model . . . . .	10
3. EXPERIMENT . . . . .	10
(a) Thermogravimetry . . . . .	13
(b) Burning Tests . . . . .	14
(c) A Note on the Heating Rates . . . . .	23
4. RESULTS AND DISCUSSION . . . . .	23
(a) Burning Test Results . . . . .	23
(b) Predictions by Thermochemical Model . . . . .	28
5. SUMMARY . . . . .	33
REFERENCES . . . . .	35

Appendix A - Derivation of  $r$

Appendix B - Computer Program

Appendix C - Some Comments on Burning Tests Procedure

Appendix D - Photographs of Equipment and Burned Samples

Appendix E - Comparison With Factory Mutual's Numerical Model

## LIST OF ILLUSTRATIONS

<u>Figures</u>	Page
1. Geometry, Considered in the First Case Where $T = T_0$ at $X = \infty$	4
2. Geometry Considered in the Second Case Where $T = T_c$ at $X = X_L$	6
3. Geometry of the Third Case, a Three-Layered System	8
4. Thermogravimetric Diagram of Seat Cushion Cover Fabric	15
5. Thermogravimetric Diagrams of Three Fire Blocking Layers, LS-200, Vonar, and Preox	16
6. Thermogravimetric Diagrams of Two Foams, Polyurethane Foam and Polyimide Foam	17
7. Arrhenius Plot of Seat Cushion Cover Fabric, Deduced From TG Shown in Figure 4	18
8. Arrhenius Plots of Two Fire Blocking Layers, Vonar and Preox, Deduced from TG Shown in Figure 5	19
9. Arrhenius Plots of Polyurethane Foam and Polyimide Foam, Deduced from TG Shown in Figure 6	20
10. Weight and Temperature Change Data Measured During the Burn Tests in a Modified NBS Smoke Density Chamber With and Without a Fire Blocking Layer	25-26
11. Temperature Profiles of Multilayered Seat Cushion System With and Without a Fire Blocking Layer	29-32
12. Comparison of Mass Losses Predicted by the Model and Experimental Data	34

## LIST OF TABLES

<u>Table</u>	<u>Page</u>
1. (a) Fundamental Properties of Seat Cushion Component Materials	21
(b) Kinetics Constants and Heat of Degradation of Subdivided Thermochemical Layers	21-22

## NOMENCLATURE

B	Pre-exponential constant for thermal degradation of polymer (sec <sup>-1</sup> )
c	Specific heat (cal gm <sup>-1</sup> °C <sup>-1</sup> )
D	Heat of degradation (cal gm <sup>-1</sup> )
E	Activation energy for thermal degradation of polymer (cal mole <sup>-1</sup> )
FS	Remaining number of monomer units of degraded polymer
k	Thermal conductivity (cal cm <sup>-1</sup> sec <sup>-1</sup> °C <sup>-1</sup> )
N	Remaining number of polymer bonds normalized with respect to the number in the unaffected state
q	Heat flux (cal cm <sup>-2</sup> sec <sup>-1</sup> )
r	Linear regression rate of polymer solid (cm sec <sup>-1</sup> )
R	Universal gas constant (cal g-mole <sup>-1</sup> K <sup>-1</sup> )
t	Time (sec)
T	Temperature (K)
w	Weight (gm)
x	1-dimensional coordinate, i.e., depth from surface (cm)
ρ	Density (gm cm <sup>-3</sup> )
λ	The normalized regression rate eigenvalue ( $\propto 1/r^2$ )

### Subscripts

1, 2, 3	Pertaining to each of multilayers
s	Front (hot side) surface of a layer
c	Rear (cold side) surface of a layer
o	Undisturbed state
L	Thickness of a layer

### Superscripts

+, - A point immediately beyond and before, respectively

## EXECUTIVE SUMMARY

The purpose of this effort was the evaluation of the fire performance of seat blocking layers for urethane seat cushions. Because urethane cushions have been demonstrated as highly flammable under aircraft post-crash fire conditions, seat blocking layers have been proposed as a method of protecting the urethane from involvement with the fire. Many hypothesis have been proposed to explain the manner in which fire blocking layers achieve their desired effect, but such hypothesis have not been verified in any quantitative sense.

A condensed phase thermochemical modeling theory previously developed for single component aircraft materials was modified to handle multi-layered materials. The model is based on analytical heat and mass transfer relationships. Required experimental inputs into the model are material thermal properties typical of heat transfer calculations. Additionally, the analysis technique known as thermogravimetric analysis is used to establish the parameters that describe the thermal breakdown of the plastic material when exposed to fire.

Parallel to this analytic work, experimental tests were conducted on sample blocking layer materials in a modified smoke chamber of National Bureau of Standards (NBS) variety. The samples were exposed to specified heating rates and both their weight loss and thermal behavior were measured. This measured behavior is compared with the predictions of the analytic evaluation.

The findings of the investigation are that the thermochemical model can predict the effectiveness of seat blocking layers within a limited range of fire exposure conditions. Additionally, the NBS Smoke Chamber is a useful small scale test for screening candidate seat blocking layers.

## 1. INTRODUCTION

Aircraft cabin fire safety has been one of the major research and development activities for the Federal Aviation Administration (FAA) for more than 30 years in the past (reference 1). The reasons for such a long-term commitment are the fatalities observed, unique environment in an aircraft (such as small space, high density of people, limited access for egress, etc.), and the complexity in fire phenomenon itself.

In general, fire potential in a passenger aircraft can be due to jet fuel and cabin interior materials. However, a fire involving only the interior materials can occur in-flight or on the ground. In the case of a postcrash fire, a jet fuel fire can ignite interior materials, and the hazards arise from interaction of the fuel fire with interior materials (reference 2).

As planned earlier (reference 4), this work is on the prediction of material burning phenomena through condensed phase thermochemical modeling, and does not involve gas dynamic modeling. These two approaches may well be bridged in the future (reference 3) with sufficient progress.

Among the cabin interior materials such as carpet, seats, window screens, sidewall panels, ceiling and partitions, one of the most flammable materials, seat cushion, is the subject of the current work. Previous work, last year, (reference 4) was on wool carpet and polyurethane foam treated as single layer materials.

In this work, the seat is treated as a system of multilayered polymeric materials, consisting of a seat cover fabric, polyurethane foam cushion, and a fire blocking layer.

In addition to analytically predicting the burning behavior of multilayered systems as a function of heat flux and layers' thicknesses, burning tests are conducted in a modified NBS density chamber to verify the temperature profile and weight losses predicted by the model.

## 2. THEORETICAL

The thermochemical model used in this work is an extension of the model developed earlier for a single-layered material. As a matter of fact, the approach used in predicting the burning rate of wool carpet with a char layer on its top (reference 4) is a particular form of such an extension. Another earlier work (reference 5) applied to the study of burning rate of composite materials is also referenced for the generalized extension.

### (a) The Applicability of the Steady-State Model

There are two aspects of this time-independent model which should be recognized. One is the coordinate transformation (a la Spalding (reference 11)) that enables us to treat the regressing surface as if it were stationary and the other is the more fundamental argument on the relative time scales.

- (1) The assumption of time-independent degradation enables one to see that  $d/dt$  can be written as  $r \cdot d/dx$ . This transformation, the authors believe, was first introduced in combustion theory by Brian Spalding (reference 11) in his treatment of laminar flame propagation in premixed gases. The point is that the "transient" propagation can be viewed as steady-state in another coordinate frame.
- (2) In any case, the assumption  $\partial/\partial t \equiv 0$  implies that the characteristic heat transfer time is small compared to the characteristic "flow" or thermal wave propagation time. The time needed for the establishment of the fully developed temperature profile in the solid is, of course, infinity from the first order nature of the equation's; recall

$$(T_x - T_0)/(T_{xs} - T_0) = 1 - \exp(-\alpha t/x^2)$$

where  $T_x$  is the temperature at any depth  $x$  (from the surface),  $T_0$  is the initial temperature,  $T_{xs}$  is the steady-state temperature at depth  $x$ ,  $\alpha$  is the thermal diffusivity

( $\equiv k/\rho c$ ),  $t$  is the time. Thus the time-independence assumption in our treatment relies on a small value for the characteristic time  $L^2/\alpha$  for its validity. In the case on hand, the following approximate numbers lend credence to this approach. Typically  $\alpha = 10^{-3} \text{ cm}^2/\text{sec}$ . The "characteristic" thermal depth, i.e., the distance from the surface to reach  $1/e$  of the full temperature difference is 0.2 cm (for example from figure 11, first plot) at 120 seconds (2 min.); or

$$t \frac{\alpha}{x^2} \approx 3.0$$

leading to

$$1 - \exp(-\alpha t/x^2) > 0.9$$

Thus the steady-state assumption is valid in our case of heating rates.

### (b) Mathematical Formulation

As in the previous work, the geometry under consideration is one-dimensional, and the steady-state condition is assumed throughout the analysis. The following mathematical analyses for a multilayered system are divided into three parts: the first part is for the case where a single-layered material is so thick that the rear surface (cold side) temperature is undisturbed; the second part is the same as above except that the rear surface temperature has been raised substantially, and the third case is the extension for a layer which is on top of the above two layers and has as many colder layers underneath it as determined by the configuration.

In all three cases, the governing equation for each layer is the same, that is,

$$k \frac{d^2T}{dx^2} + \rho c r \frac{dT}{dx} = D \rho N B \exp (-E/RT) \quad (1)$$

The right-hand side of this equation is the heat sink term corresponding to the first-order degradation reaction described by

$$-\frac{1}{N} \frac{dN}{dt} = B \exp (-E/RT) \quad (2)$$

The symbol  $N$  represents the number of remaining bonds per unit mass, normalized with respect to that of fresh, unburned material. The symbol  $D$ , the heat of degradation, is "positive" when the degradation reaction is endothermic and "negative" when exothermic.

The boundary conditions, however, are different for each case.

(1) First case, i.e.,  $T = T_0$  at  $x = \infty$

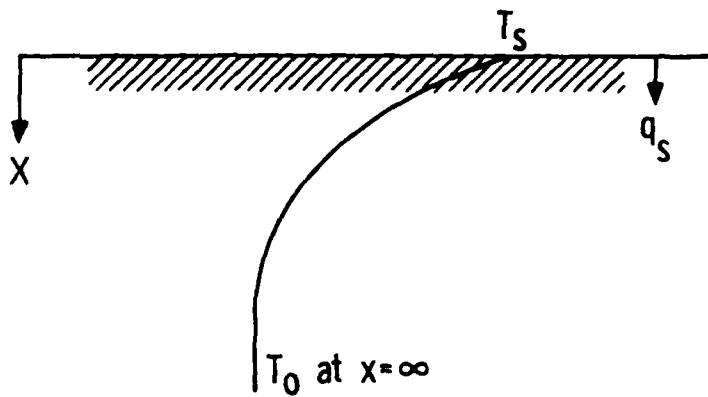


Figure 1. Geometry considered in the first case where  $T = T_0$  at  $x = \infty$

When the rear surface (cold side) temperature is undisturbed, the boundary conditions for equations (2) and (3) are, respectively,

$$\begin{aligned} T &= T_s & \text{at } x = 0 \\ T &= T_0 & \text{at } x = \infty \end{aligned} \quad (3)$$

$$\begin{aligned} N &= 1 - \frac{1}{FS} & \text{at } x = 0 \\ N &= 1 & \text{at } x = \infty \end{aligned} \quad (4)$$

The symbol FS stands for fragment size, an average number of monomer units of degraded polymer.

Assuming constant values of material properties, the solution for  $r$ , surface regression rate, has been obtained (reference 6) by singular perturbation methods with matching the solution for the inner (surface) and outer (deep solid) regions. For the steady state system,  $r$  is given as:

$$r = \sqrt{\frac{(k/\rho c) B \exp(-E/RT_s)}{\left(\frac{E}{RT_s}\right)\left(\frac{T_s - T_0}{T_s}\right) \left[ \left\{1 + \frac{D}{c(T_s - T_0)}\right\} \ln\left(\frac{FS}{FS-1}\right) - \frac{D}{c(T_s - T_0)FS} \right]}} \quad (5)$$

And the integration of equation (1) will show that the conductive heat flux at the hot surface is

$$q_s = \rho r [c(T_s - T_0) + D/FS] \quad (6)$$

if there is zero heat flux at the rear surface. Thus, all the thermal energy flowing into the layer can be said to be totally consumed within the layer.

(2) Second Case, i.e.,  $T = T_C (> T_0)$  at  $x > 0$

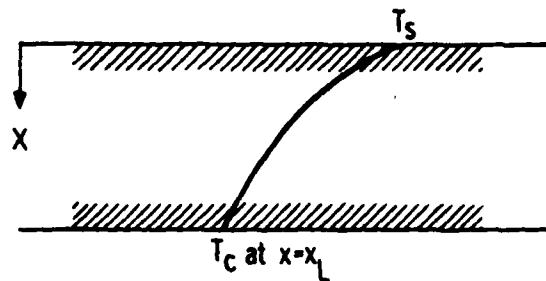


Figure 2. Geometry considered in the second case where  $T = T_C$  at  $x = x_L$

This is the case when the front (hot) and rear surface temperatures of a layer are substantially higher than the unperturbed condition of  $T_0$ . The governing equations are the same as above, i.e., equations (1), (2), and the boundary conditions are

$$\begin{aligned} T &= T_S & \text{at } x = 0 \\ T &= T_C & \text{at } x = x_L \end{aligned} \quad (7)$$

$$\begin{aligned} N &= 1 - \frac{1}{FS_S} & \text{at } x = 0 \\ N &= 1 - \frac{1}{FS_C} & \text{at } x = x_L \end{aligned} \quad (8)$$

Assuming that  $T_C$  is close to  $T_S$ , that is

$$\frac{T_S - T_C}{T_S - T_0} < 0.1 \quad (9)$$

the solution for  $r$  is given by

$$r = \sqrt{\frac{(k/\rho c) B \exp(-E/RT_S)}{\lambda \left(\frac{E}{RT_S}\right) \left(\frac{T_S - T_0}{T_S}\right)}} \quad (10)$$

The details of this derivation is shown in appendix A.

The integration of equation (1) in the region between the front and rear surfaces will give the net heat flux, that is, the incoming heat flux at the front surface minus the one at the rear surface.

$$q_{\text{net}} = \rho r \left[ c (T_s - T_c) + D \left\{ \frac{1}{FS_s} - \frac{1}{FS_c} \right\} \right] \quad (11)$$

In other words, the above equation determines the net heat consumed within the layer volume, as sensible heat and heat of degradation.

### (3) Third Case of a multilayered system

In the above two cases, the analyses are for a homogeneous single layer for which thermal, physical and chemical properties are constant and uniform in every part of the layer. If a system consists of more than one material, like the example of a cover fabric - blocking layer - foam, those properties are widely different and hence will be treated as a multilayered system.

Strictly speaking, even a single component material should be treated as a multilayered system, if the temperature drop from the front to the rear surface is so large that the thermal, physical and chemical properties can not be considered constant and uniform within the layer.

In other words, when a material of certain thickness is burning, it is a multilayered system from the view point of thermochemical behavior. As shown later, B and E of any single material employed in this study do not remain constant over the temperature range of interest but vary considerably.

For this reason, each component material of a multilayered seat cushion system is in itself considered multilayered, divided by temperatures at which any of the thermal and/or thermochemical properties changes substantially. Hence, the total number of layers, i.e., thermochemical layers, of a seat cushion system become equal to the sum of the thermochemical layers of each component material.

For the ease of analysis, first suppose that a system of three layers is undergoing pyrolysis reaction, as depicted in figure 3. The same analysis can be extended to any system with more than three layers with the change in subscripts.

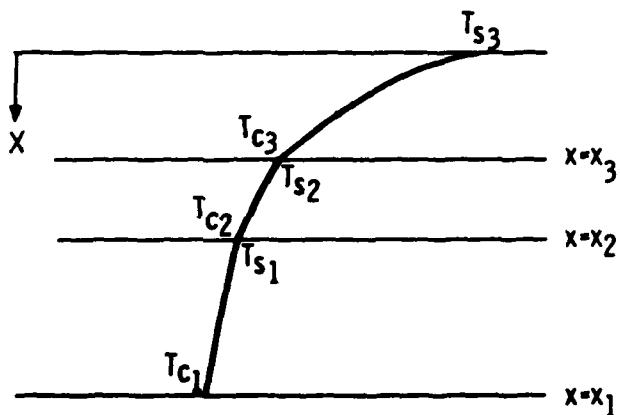


Figure 3. Geometry of the third case, a three-layered system

The conductive heat flux at the top surface ( $x=0$ ) should be the sum of all the energy, i.e., the sensible heat and the heat of degradation, required for the pyrolysis of all three layers in this 1-D approximation. The energy required for each individual layer is shown in equation (6) or equation (11). Then, the incoming heat flux at the top surface of the first, second, and the third layer is, respectively,

$$\begin{aligned}
 -k_3 \frac{dT}{dx} \Big|_{x=0^+} &= \rho_3 r_3 \left[ c_3 (T_{s_3} - T_{c_3}) + D_3 \left( \frac{1}{FS_{s_3}} - \frac{1}{FS_{c_3}} \right) \right] \\
 &+ \rho_2 r_2 \left[ c_2 (T_{s_2} - T_{c_2}) + D_2 \left( \frac{1}{FS_{s_2}} - \frac{1}{FS_{c_2}} \right) \right] \quad (12) \\
 &+ \rho_1 r_1 \left[ c_1 (T_{s_1} - T_{c_1}) + D_1 \left( \frac{1}{FS_{s_1}} - \frac{1}{FS_{c_1}} \right) \right]
 \end{aligned}$$

$$\begin{aligned}
 -k_2 \frac{dT}{dx} \Big|_{x=x_3^+} &= \rho_2 r_2 \left[ c_2 (T_{s_2} - T_{c_2}) + D_2 \left( \frac{1}{FS_{s_2}} - \frac{1}{FS_{c_2}} \right) \right] \\
 &+ \rho_1 r_1 \left[ c_1 (T_{s_1} - T_{c_1}) + D_1 \left( \frac{1}{FS_{s_1}} - \frac{1}{FS_{c_1}} \right) \right] \quad (13)
 \end{aligned}$$

and

$$-k_1 \frac{dT}{dx} \Big|_{x=x_2^+} = \rho_1 r_1 \left[ c_1 (T_{s_1} - T_{c_1}) + D_1 \left( \frac{1}{FS_{s_1}} - \frac{1}{FS_{c_1}} \right) \right] \quad (14)$$

There are however, two kinds of physical constraints applied at each interface. The first kind is that

$$\begin{aligned}
 T_{c_3} &= T_{s_2} \\
 T_{c_2} &= T_{s_1}
 \end{aligned} \quad (15)$$

and the second is that

$$\begin{aligned}
 -k_3 \frac{dT}{dx} \Big|_{x=x_3^-} &= -k_2 \frac{dT}{dx} \Big|_{x=x_3^+} \\
 -k_2 \frac{dT}{dx} \Big|_{x=x_2^-} &= -k_1 \frac{dT}{dx} \Big|_{x=x_2^+}
 \end{aligned} \quad (16)$$

The above two conditions are to satisfy the continuity of temperature and heat flux at interfaces.

In the expression of equations (12), (13), or (14), each  $r$  is implicitly defined. In other words, the heat fluxes at the upper and lower boundaries of each layer are used in obtaining the  $r$  of that layer (as shown in appendix A), and are in turn used to define the heat fluxes [as shown in equations (12), (13), and (14)]. This poses no problem in actual calculations since any iterative method leads to fairly rapidly converging answers. The significance of this fact is, however, that the calculation of  $r$ 's should proceed layer by layer, from the rear (coldest) to the front (hottest), or from the front to the rear.

(c) Computer Program

Based on the theory discussed above, two computer programs are developed and listed in appendix B. The first program, (Computer Program A) takes the interfacial temperatures as the input data, and the second (Computer Program B) takes the temperature and the conductive heat flux at the front surface as its input. The choice among these two programs is dictated firstly by the available boundary conditions, and the other may be used as a complementary to obtain additional information.

(d) The Overall Logic and Characteristics of the Model

The model was originally developed to account for the host of inconsistencies in polymer degradation and burning rate data. The most important distinguishing characteristics are the recognition that the mean molecular weight of the vaporizing molecules need not be equal to the monomer molecular weight. Instead, a vapor pressure equilibrium criterion was used to unambiguously specify the mean molecular weight at the surface. This procedure not only removed an

annoying arbitrariness at the surface, but also checked well, in terms of results, with data from various classes of polymer combustion. Under the assumption of first order Arrhenius kinetics for the degradation of the subsurface polymer, the model, in its present state, cannot handle cases where extensive degradation beyond the monomer stage takes place before vaporization. This is not as severe a limitation as it may seem. It has been documented (e.g., see Stanley Martin, X Symposium (International) on Combustion) that many of the degradation reactions actually take place after the major precursors (i.e., products of partial degradation) have left the surface.

The model also assumes that the thermal wave front moves at a uniform speed in the solid. This thermal wave can be associated with the mean surface (and hence the wave speed is indeed the burning rate or "regression rate") only when  $\Delta_i$  of the solid (condensed phase) gets transformed into vapor. This is indeed the case in the burning of simple plastics (e.g., plexiglas, polyethylene, polyurethane . . . ). In the case on hand, substantial charring occurs indicating that the familiar "regression rate" needs careful interpretation. For example, it is a familiar fact that in wood burning the reactive portion (cellulose and hemicellulose) leaves the solid framework of lignin char. In such cases of charring solids, the regression rate is more appropriately associated with the velocity of the thermal wave in the wake of which the reactive portion gets vaporized and leaves the char. Thus, it may appear that the surface is not really regressing in the physical sense, although the reactive portion is. Up to this point, the interpretation of the regression rate of this model is clear. The interpretation gets more complex as multilayered materials are concerned. Hence, clearly, the regression rate of the **II** layer, if interpreted literally, will give rise to a "void" or separation between the bottom of the I layer and the initial position of the **II** layer top surface. However, the complexity is removed when one recognizes that the regression meant here is really the movement of the thermal wave in the solid layers while the char framework (end

product of the degradation) is stationary in the laboratory frame of reference. The appropriate way of handling this situation mathematically is to keep the density of the material  $\rho$  as a variable. For example,  $\rho_{final}$  would not equal zero but would equal the char density which itself is equal to the  $(\rho_{initial} - \rho_{reactive})$

With this interpretation, the regression rate  $r$  is really always associated with the mass flux  $\rho r$  where  $\rho$  is not the density of the solid but is only the density of the reactive portion of the solid. However, in the evaluation of the properties such as the thermal diffusivity of the reacting solid, the full density has to be used. Also, the heat transfer through the char is characterized by a thermal conductivity coefficient of the char, and the volatiles convection (flow) through the char. This has been highly simplified in the present analysis which considers the temperature to be constant (and equal to the temperature of the bottom surface of the immediate layer above). Clearly, more work is needed to mathematically incorporate into the model, the realistic char formation, flow and heat transfer.

The "void" that is mentioned in the experimental portion is not the same on this void or separation. Experimentally it is found that the mechanical movement of the volatiles frequently causes a void in the assembly and physical separation of layers. This is a very complex process mathematically.

Considering all of these complexities, the agreement between the theory and experiments in this study is thought to be encouraging.

### 3. EXPERIMENTS

The experiments of this work consist of two parts; one is to obtain the kinetics constants using thermogravimetry (TG) and the other is, through actual burning tests, to determine the temperature profile established within a seat cushion assembly and the weight loss during pyrolysis. The experimental results are compared with the model predictions.

### (a) Thermogravimetry

This experiment heats up a small sample suspended inside a furnace and records the weight change as a function of temperature. The kinetics constants, B and E, are deduced then from an Arrhenius plot, a plot of  $\log \{(1/w) \cdot -(dw/dt)\}$  versus reciprocal of absolute temperature  $(1/T)$ .

It was known from previous work (reference 4) that the ambient gas composition, more precisely the oxygen concentration, affects the kinetics constants. This is a very important point. It has long been recognized that even minor ( $\sim 1\%$ ) concentrations of certain oxidative species can significantly alter (by almost an order-of-magnitude) the degradation rates of polymers. And yet no detailed study seems to be available in this area. One study that addressed this question specifically stopped short of actually demonstrating the effects but had to make valid (but indirect) deductive arguments to point the importance. In any combustion situation, especially with flow of gases over the surface (a very common aircraft fire scenario), the oxygen concentration at the burning surface appears to be around 0.1%-1% (Wooldridge and Muzzy (reference 8), Kulgein (reference 9), Fennimore and Jones (reference 10)). This measured non-zero concentration is significant, because earlier JPL work under FAA sponsorship (reference 4) indicated that actual aircraft interior materials exhibited a strong dependence of the kinetics constants of degradation on small concentrations of oxygen in a stream of oxygen in an inert. When it is realized that turbulent transport offers a mechanism for the availability of small concentrations of oxygen at the burning surface, it is easy to see that these oxygen concentration effects could be important in predicting full-scale burning behavior. Full-scale diffusion flames would be large enough to be fully turbulent.

Nevertheless, all the present TG experiments are performed in pure nitrogen except for seat cover fabric, due to the limited scope of the work.

The TG diagrams of the cover fabric, fire blocking layers (Vonar®, LS-200, and Preox®), and foams (polyurethane and polyimide) are shown in figures 4, 5, and 6, respectively. The Arrhenius plots of figures 7, 8, and 9 show that the pyrolysis of each of these materials cannot be described as a single first-order reaction. Thus, the entire reaction temperature range is subdivided into multiple segments in each of which the reaction rate is reasonably accurately described by the first-order Arrhenius expression. Figure 7 shows how it is done for the cover fabric, as an example, with six subdivided segments. Each segment comprises one thermochemical layer as discussed in the previous section.

Kinetics constants obtained along with other thermal and physical data of the materials used in this study are shown in Table I. These are used as input data for the computer program shown in Appendix B.

#### (b) Burning Tests

In order to measure the temperature profile and weight loss of samples, a modified NBS Smoke Density Chamber is used. The radiative heat flux was provided by a high heat flux furnace (Mellen furnace Model No. 10, with the maximum heat flux of  $12 \text{ W/cm}^2$  or  $10.6 \text{ BTU/ft}^2 \text{ sec}$ ). The originally equipped furnace for the NBS chamber could provide  $2.5 \text{ W/cm}^2$ . For the temperature distribution measurement, nine thermocouples (Pt vs. Pt-10% Rh) were connected to a multichannel recorder (Leeds and Northrup's Speedomax Model 251, 12 channel, with resolution time of 1 second for each reading). The weight measuring device was a transducer-type cantilever beam, connected to an amplifier and then to a strip-chart recorder.

The furnace is heated up to, and maintained at,  $850^\circ\text{C}$  during the tests. After the furnace has reached a steady state, a radiation shield is removed and the seat cushion is exposed directly to the furnace. The radiative heat flux is around  $\sim 1.6 \text{ BTu/ft}^2 \text{ sec}$ .

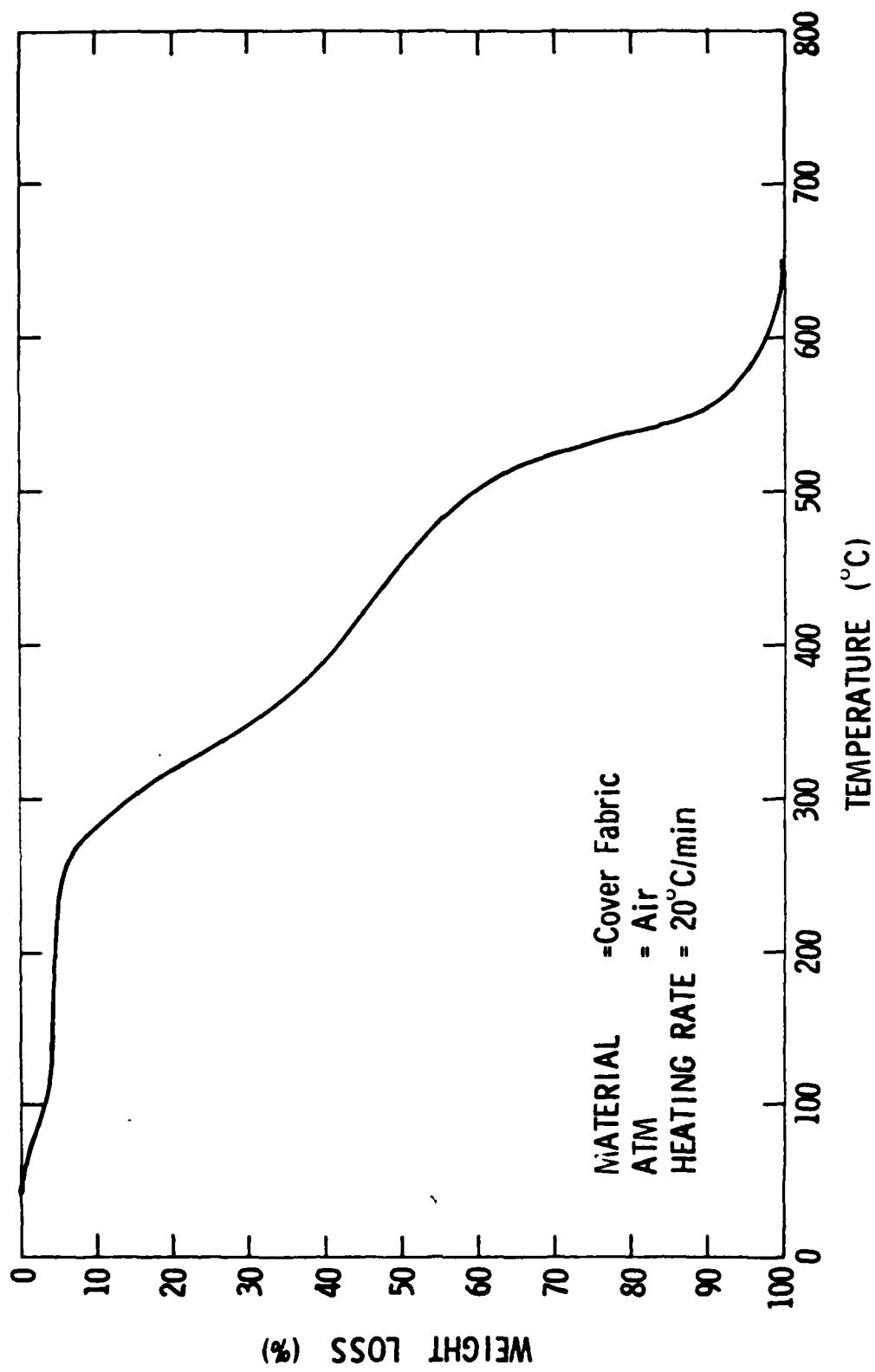


FIGURE 4. THERMOGRAVIMETRIC DIAGRAM OF SEAT CUSHION COVER FABRIC.

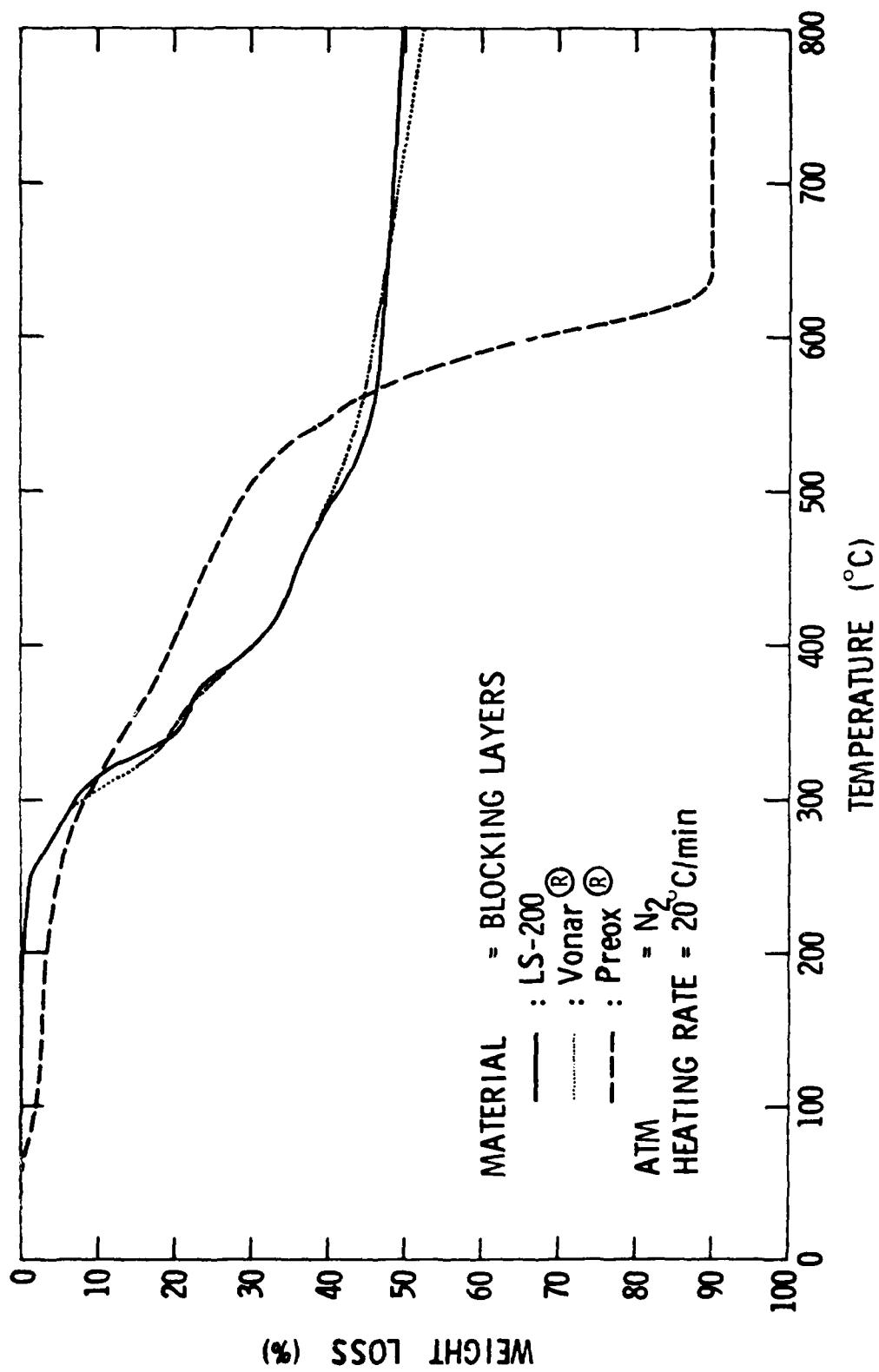


FIGURE 5. THERMOGRAVIMETRIC DIAGRAMS OF THREE FIRE BLOCKING LAYERS, LS-200, VONAR, AND PREOX.

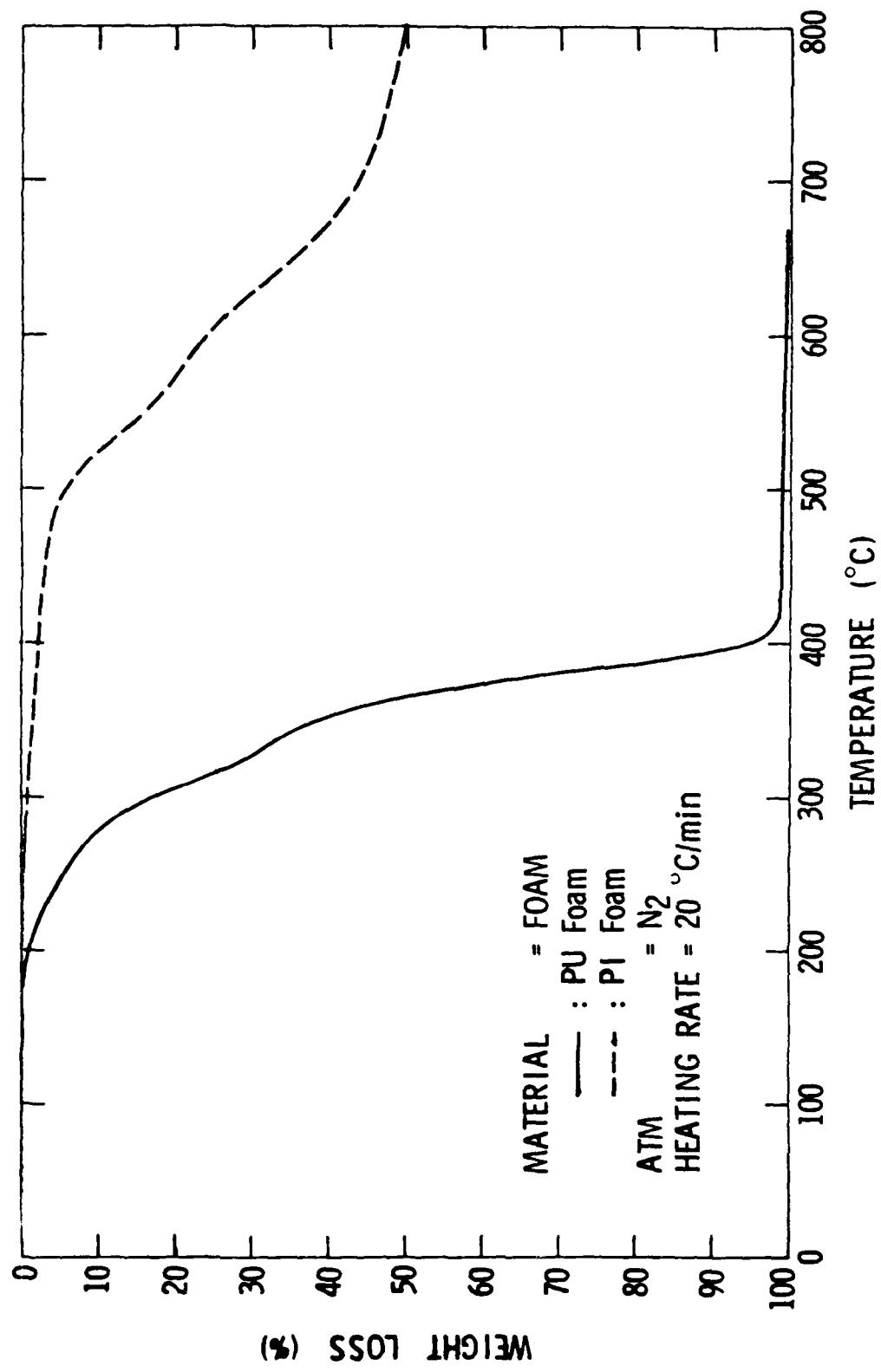


FIGURE 6. THERMOGRAVIMETRIC DIAGRAMS OF TWO FOAMS, POLYURETHANE FOAM AND POLYIMIDE FOAM.

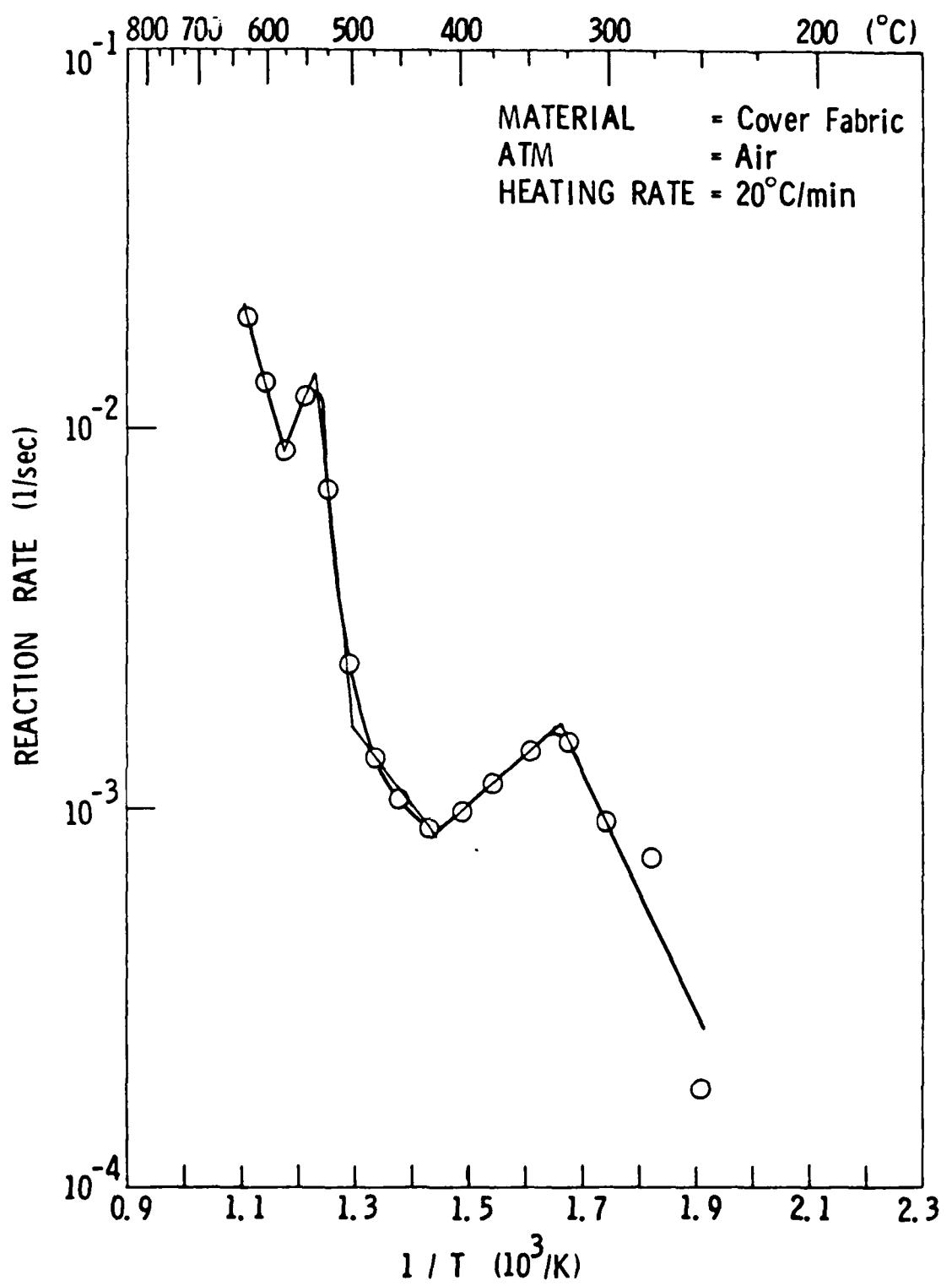


FIGURE 7. ARRHENIUS PLOT OF SEAT CUSHION COVER FABRIC, DEDUCED FROM TG SHOWN IN FIGURE 4.

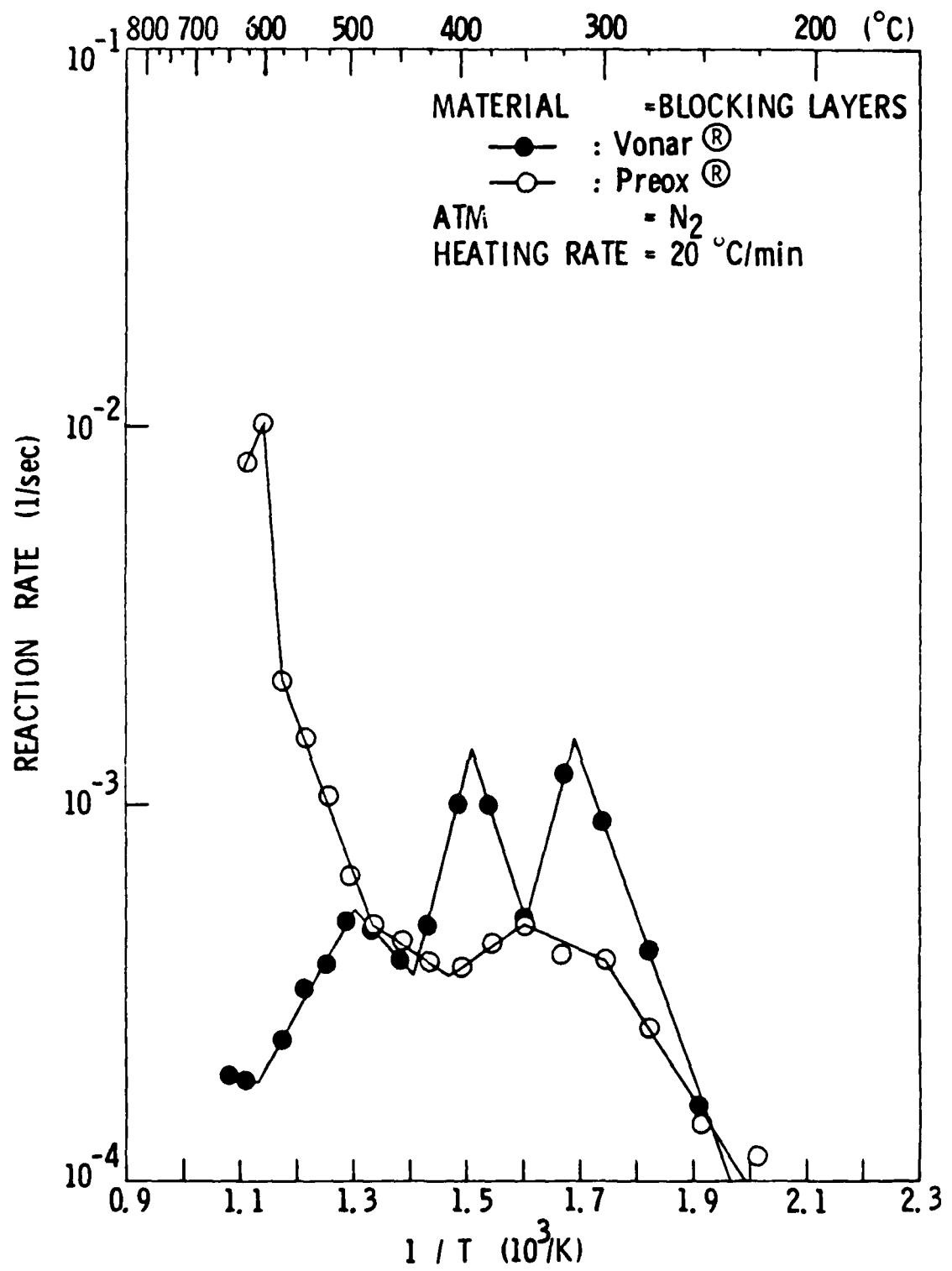


FIGURE 8. ARRHENIUS PLOTS OF TWO FIRE BLOCKING LAYERS, VONAR AND PREOX,  
DEDUCED FROM TG SHOWN IN FIGURE 5.

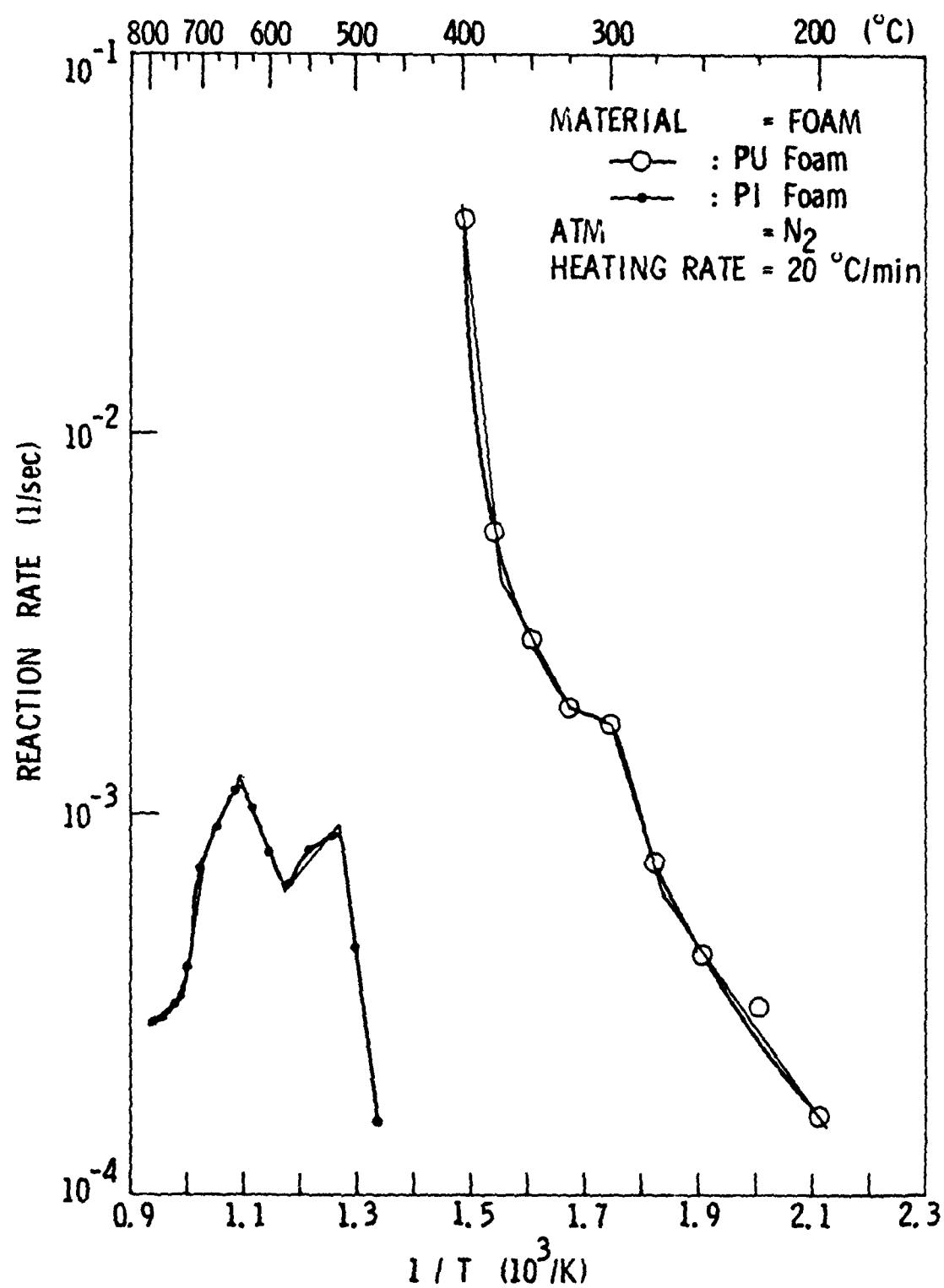


FIGURE 9. ARRHENIUS PLOTS OF POLYURETHANE FOAM AND POLYIMIDE FOAM,  
DEDUCED FROM TG SHOWN IN FIGURE 6.

Table I(a). Fundamental Properties of Seat Cushion Component Material

	Density g cm <sup>3</sup>	c Specific Heat cal g.°C	k* Thermal Conductivity cal cm. sec. °C	x <sub>L</sub> Thickness cm
<u>Seat Cover Fabric</u>	0.4	0.3	0.0001	0.1
<u>Fire Blocking Layer</u>				
LS-200	0.12	0.3	0.0002	1.2
Vonar®	0.146	0.345	0.0002	0.8
Preox®	0.62	0.3	0.00034	0.1
<u>Foam</u>				
Polyurethane	0.03	0.4	0.00010~0.00034	10.6
Polyimide	0.023	0.2	0.0001	-

(b) Kinetics Constants and Heat of Degradation of Subdivided Thermochemical Layers

	T Tempera- ture Range (K)	B Pre Exponential Factor (1/sec)	E Activation Energy (cal/mole)	D Heat of Degrada- tion (cal/g)
Seat Cover Fabric	848-898 813-848 763-813 689-763 602-689 523-602	2.13 x 10 <sup>4</sup> 5.06 x 10 <sup>-8</sup> 2.24 x 10 <sup>13</sup> 2.51 x 10 <sup>-1</sup> 7.30 x 10 <sup>-6</sup> 1.57 x 10 <sup>2</sup>	24800 -20300 56500 7800 -6500 13700	-2.9 -2.9 -2.9 -2.3 -16. -16.3

\* Source: NASA Ames Research Center -- Handbook of Chemistry and Physics

**(b) Kinetics Constants and Heat of Degradation  
of Subdivided Thermochemical Layers (Cont'd)**

	T Tempera- ture Range (K)	B	E	D
		Pre Exponential Factor (1/sec)	Activation Energy (cal/mole)	Heat of Degrada- tion (cal/g)
LS-200	877-923	$1.17 \times 10^{-5}$	-3857	0.5
	767-877	$7.17 \times 10^{-11}$	-24770	9.6
	723-767	$5.17 \times 10^1$	16818	-8.4
	698-723	$1.04 \times 10^{-4}$	-2020	-19.1
	673-698	$8.28 \times 10^{-14}$	-31083	-16.7
	648-673	$6.56 \times 10^{-3}$	2475	-8.4
	623-648	$2.46 \times 10^6$	27897	-20.3
	598-623	$1.61 \times 10^{-17}$	-38190	7.9
	573-598	$2.11 \times 10^7$	27800	71.9
	548-573	$2.01 \times 10^{-1}$	6772	24.9
	473-548	$7.83 \times 10^3$	18279	54.9
Vonar <sup>®</sup>	881-923	$3.37 \times 10^{-4}$	1043	1.0
	766-881	$1.66 \times 10^{-7}$	-12285	9.6
	709-766	$8.94 \times 10^{-2}$	7800	-23.9
	661-709	$2.37 \times 10^{-12}$	-26532	-28.7
	623-661	$2.75 \times 10^4$	22078	-28.7
	591-623	$3.94 \times 10^{-13}$	-25933	50.9
	473-591	$2.33 \times 10^4$	19435	107.5
Preox <sup>®</sup>	873-908	$5.42 \times 10^{-6}$	-13061	0.5
	848-873	$8.55 \times 10^{20}$	91578	0.5
	748-848	$1.55 \times 10^2$	18860	0.5
	683-748	$9.85 \times 10^{-3}$	4500	-13.9
	623-683	$1.43 \times 10^{-5}$	-4365	-13.9
	573-623	$5.87 \times 10^{-3}$	3085	-13.9
	498-573	$5.94 \times 10^0$	10963	0.5
Polyurethane Foam	644-688	$8.44 \times 10^{19}$	65760	3.1
	598-644	$6.17 \times 10^1$	12335	11.9
	573-598	$1.77 \times 10^{-2}$	2646	2.5
	544-573	$2.60 \times 10^5$	21435	2.5
	448-544	$6.00 \times 10^0$	9900	0.5
Polyimide Foam	1006-1073	$3.06 \times 10^{-5}$	-4763	0.5
	973-1006	$3.12 \times 10^{-14}$	-46170	0.5
	915-973	$1.80 \times 10^{-7}$	-16070	0.5
	854-915	$1.33 \times 10^1$	16860	-7.3
	796-854	$6.11 \times 10^{-6}$	-7890	-7.3
	711-796	$1.07 \times 10^7$	36705	-7.0
	673-711	$7.63 \times 10^{-3}$	6938	-0.3

(c) A Note on the Heating Rates

The actual heat fluxes encountered in aircraft crash situations can approach  $10 \text{ W/cm}^2$  -- a fact that led investigators to recognize that the NBS smoke-density chamber, with its original  $2.5 \text{ W/m}^2$  furnace, may have to be modified for realistic simulations. In the present effort, the heat fluxes meant for the FAA samples were planned to be gradually increased from a low value ( $1 \text{ W/cm}^2$ ) to the full value of  $6-10 \text{ W/cm}^2$ . It was the plan to verify the model at low heat fluxes first in a logical sequence of increasing heat fluxes. Thus, at the time of the present reporting the lower heat fluxes have been investigated. The capability exists for the higher heat fluxes. However, the actual tests await future research support.

Since the initial setup of equipment and trial runs, several modifications and improvements have been made on the sample preparation and run procedure, in order to obtain meaningful and reproducible test results. These are listed in appendix C for future reference.

A pair of thermocouples are placed at every interface, including the front surface. And additional five are placed within the polyurethane foam, 0.5 inches apart from each other along the axis. A test is always run in duplicate; one to measure the weight loss and the other for the temperature profile. This is because the thermocouples have such stiffness that their use would not allow free movement of the cantilever beam and the accurate measurement of weight change.

**4. RESULTS AND DISCUSSION**

(a) Burning Test Results

The pictures of the burned samples and the equipment used in this work are shown in appendix D for visual examination. The results of the burning tests in the modified NBS Smoke Density Chamber are

shown in figure 10. These are the temperature changes obtained from the multiple thermocouple readings placed at different positions within the seat cushion assemblies and the weight change with time, for the runs with and without the fire blocking layers.

The weight change data show that the sample without a fire blocking layer suffered the largest mass loss, and that the effectiveness of a fire blocking layer, if judged from the mass loss data, can be rated in increasing effectiveness on Preox®, Vonar®, and LS-200. The same order is observed when the burnt samples were examined visually.

The thermocouple reading at the sample surface facing the furnace, however, should be corrected for the caused by radiation and convection. The energy balance for the thermocouple bead can be shown as

$$h_{rad} (T_{furnace} - T_{bead}) = h_{conv} (T_{bead} - T_{gas\ film})$$

where  $h_{rad}$  and  $h_{conv}$  are the radiative and convective heat transfer coefficient, respectively, and  $T_{furnace}$ ,  $T_{bead}$ ,  $T_{gas\ film}$  are the temperatures of the furnace, thermocouple bead, and film stagnant gas film surrounding the bead, respectively. The estimated error ( $T_{actual} - T_{bead}$ ) is in the range of 25°C to 35°C under the current experimental conditions. Further detailed error analysis is not attempted here due to the lack of precise information on emittance, sample surface condition, and thermal contact between the bead and fabric fibers, which are necessary to estimate the heat transfer coefficients with sufficient precision.

Other than this, the thermocouple data lead to a few important observations.

- (1) The first one is that the front surface temperature measured by the thermocouple No. 1 reaches a steady state in about three

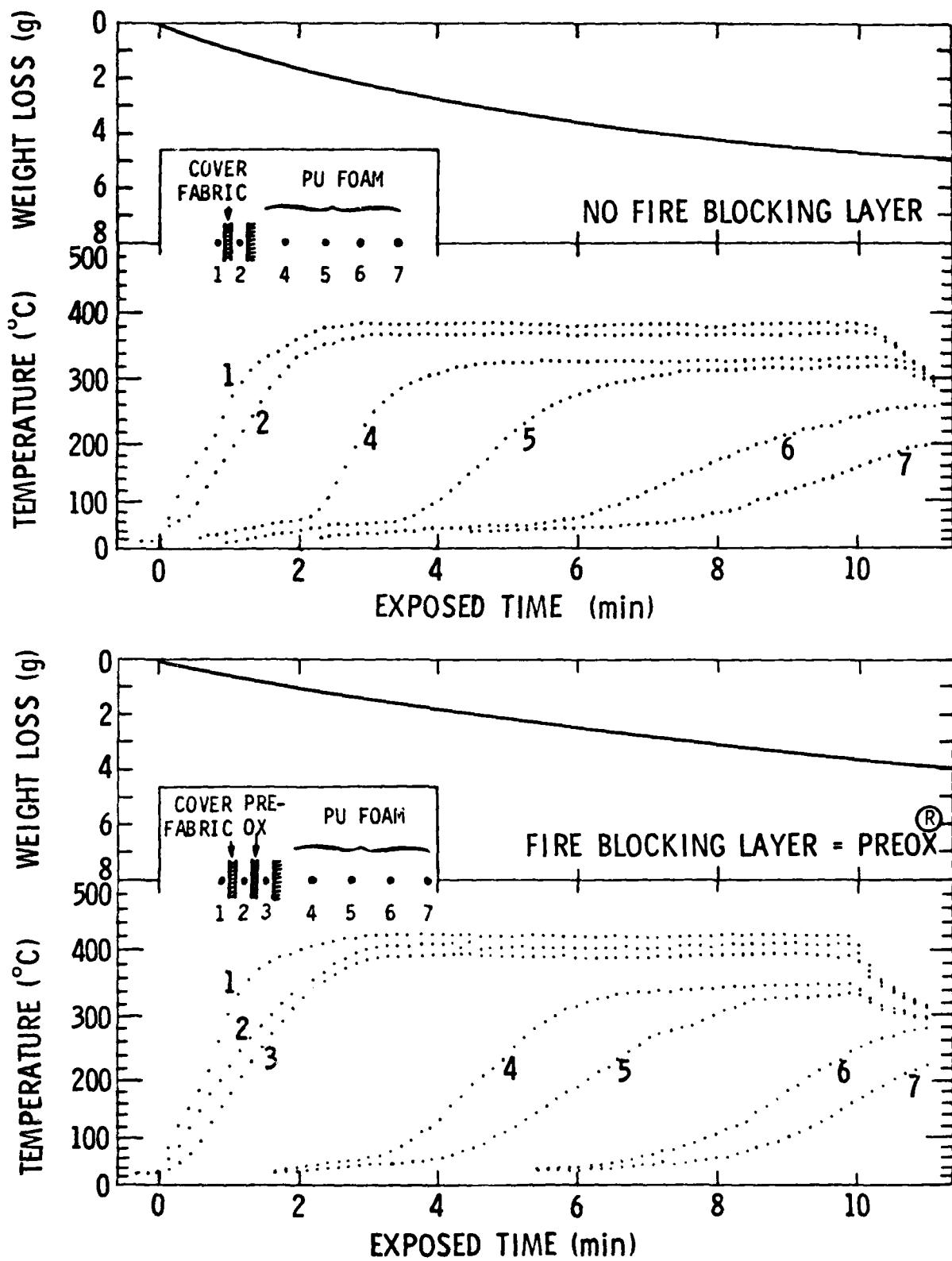


FIGURE 10. WEIGHT AND TEMPERATURE CHANGE DATA MEASURED DURING THE BURN TESTS IN A MODIFIED NBS SMOKE CHAMBER WITH AND WITHOUT A FIRE BLOCKING LAYER.  
(1 of 2 pages)

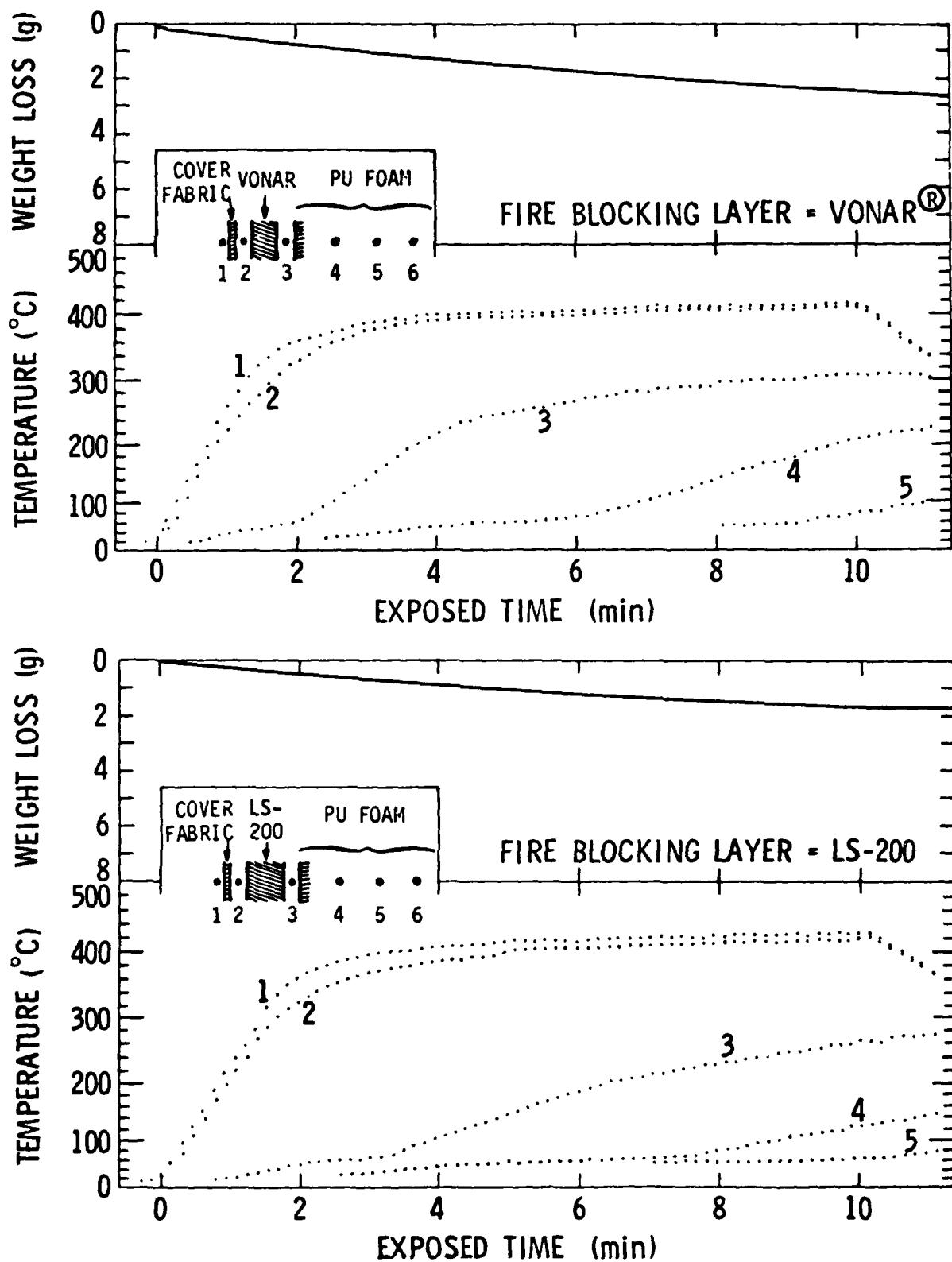


FIGURE 10.  
FIGURE 10. WEIGHT AND TEMPERATURE CHANGE DATA MEASURED DURING THE BURN  
TESTS IN A MODIFIED NBS SMOKE CHAMBER WITH AND WITHOUT A  
FIRE BLOCKING LAYER (2 of 2 pages)

minutes, and then registers essentially the same temperature thereafter. When a fire blocking layer of Vonar® or LS-200 is used, however, a very slight increase of the front surface temperature is observed. The temperature just behind the cover fabric, measured by the thermocouple No. 2, trails that of the front surface by about 15°C.

- (2) There is a further temperature drop across a fire blocking layer. This temperature drop is indicated by the vertical distance between the thermocouples No. 2 and No. 3 readings, as shown in figure 10. In the figure, LS-200 allows the largest temperature drop, Vonar® medium, and Preox® the least. Incidentally this is exactly the opposite order as the one observed in total mass loss.
- (3) The temperature-rise histories within the polyurethane foam, measured by thermocouples No. 4, 5, 6, and 7, hardly represent steady-states. Most of the measured temperatures simply keep rising steadily, as shown in figure 10.

In some cases, however, a plateau seems to appear after such monotonic increase in temperature, implying a steady-state condition attained within the foam. This behavior is believed to be caused by the formation and existence of a void or dome filled with relatively hot pyrolysis gas products in a convective flow motion. This explains why such a plateau shows up and why then these plateaus are close to each other - in other words, the temperature gradient is small. It is particularly serious when no fire blocking layer is used, because under the condition of high heat flux, the polyurethane foam liquefies and drips before complete burnout occurs. This eventually leaves an ever-growing void filled with the hot pyrolysis gas product. The hot gas of the void may flow freely and contribute to the rapid temperature rise at the rear surface.

(4) The use of a fire blocking layer helps maintain lower temperatures at the rear (colder) side essentially by reducing the heat transfer. This can be seen easily by comparing the temperatures measured by the thermocouples with the same number. For example, after 6 minutes of exposure, the thermocouple No. 5 reaches 321 K with LS-200, 323 K with Vonar<sup>®</sup>, and 463 K with Preox<sup>®</sup>, while it reaches 546 K without a fire blocking layer.

There is, however, a compensating effect near the front (hot) surface. That is, the front attains higher temperatures with the fire blocking layer than would be observed without the fire blocking layer. Figure 10 shows the front surface temperature is 701 K with LS-200, 695 K with Vonar<sup>®</sup>, 691 K with Preox<sup>®</sup>, and 643 K without one, respectively, after 10 minutes exposure. This leads to an observation that a fire blocking layer functions as if it is a reflector for heat flux.

This results in lowering the rear side temperature while maintaining the higher front side temperature.

(b) Predictions by Thermochemical Model

The temperature profiles obtained by the thermochemical model, in comparison with the experimental data, are shown in figure 11. The profile for each configuration (with and without a fire blocking layer) is calculated every 2 minutes using the Computer Program B of appendix B. (Note: The conductive heat flux at the front surface, needed as the input for the Computer Program B, was the one predicted by Computer Program A of appendix B.) In general, the calculated results are in good agreement with the experimental data. The difference is often less than 10°C, with a few exceptions of 25°C as the maximum difference. However, the predicted temperatures within the polyurethane foam show significant deviations from the measurements. The negative deviations, i.e., temperatures measured lower than predicted, are seen when the exposed time is small and/or

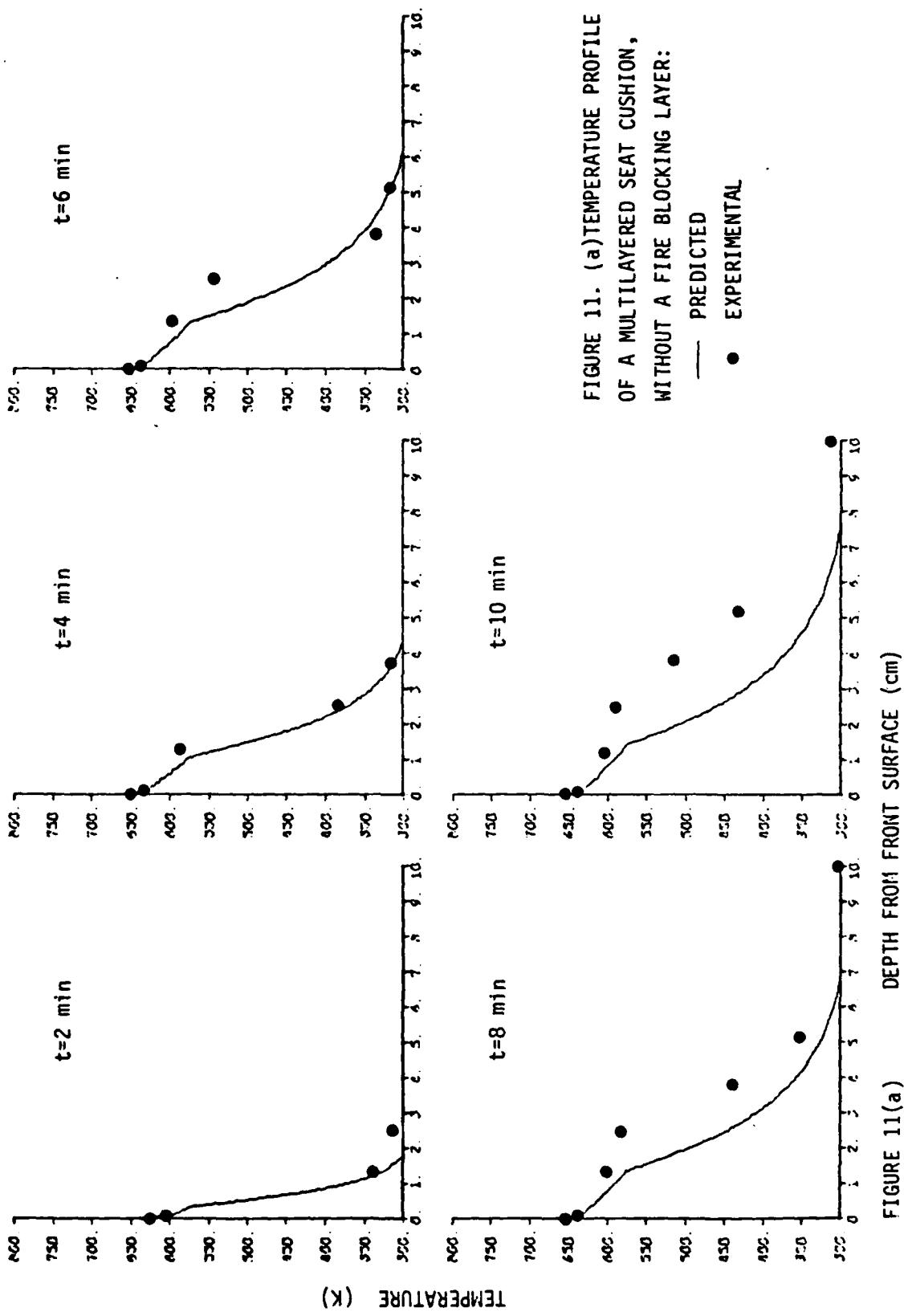


FIGURE 11(a) DEPTH FROM FRONT SURFACE (cm)

FIGURE 11. (a) TEMPERATURE PROFILE OF A MULTILAYERED SEAT CUSHION, WITHOUT A FIRE BLOCKING LAYER:  
 — PREDICTED  
 ● EXPERIMENTAL

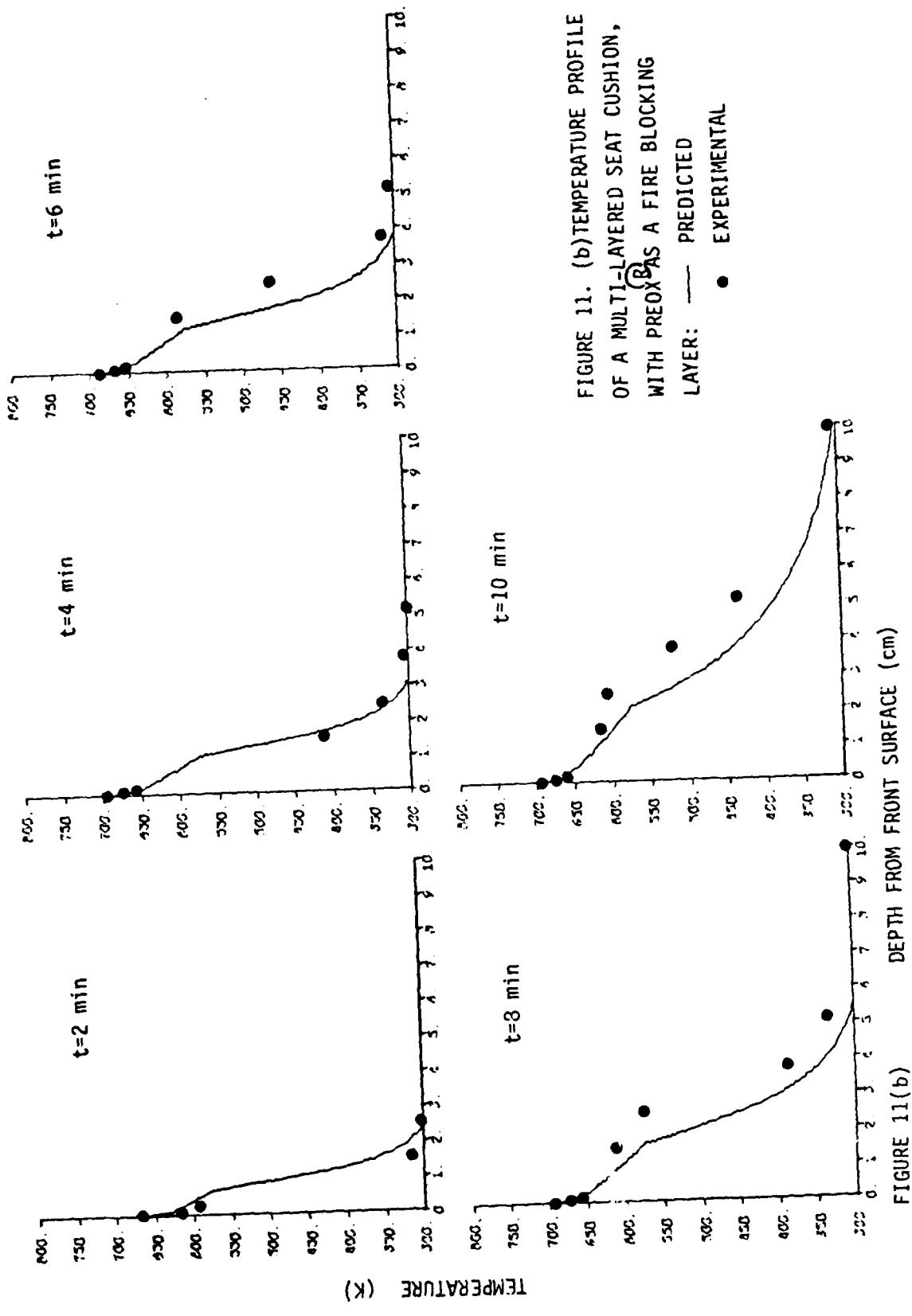


FIGURE 11(b) DEPTH FROM FRONT SURFACE (cm)

FIGURE 11. (b) TEMPERATURE PROFILE  
 OF A MULTI-LAYERED SEAT CUSHION,  
 WITH PREOX® AS A FIRE BLOCKING  
 LAYER: — PREDICTED  
 ● EXPERIMENTAL

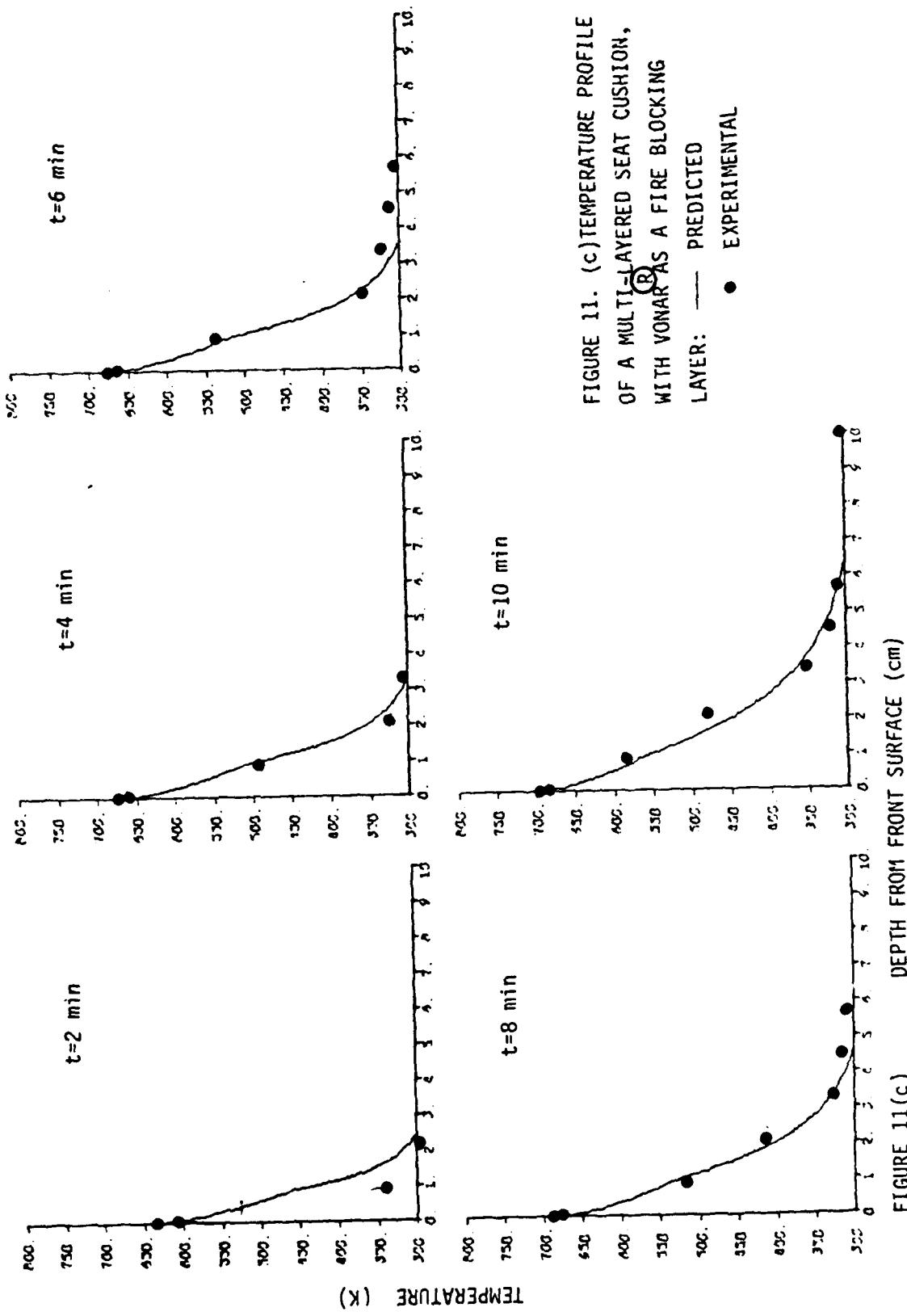


FIGURE 11. (c) TEMPERATURE PROFILE  
 OF A MULTI-LAYERED SEAT CUSHION,  
 WITH VONAR AS A FIRE BLOCKING  
 LAYER: — PREDICTED  
 ● EXPERIMENTAL

FIGURE 11(c) DEPTH FROM FRONT SURFACE (cm)

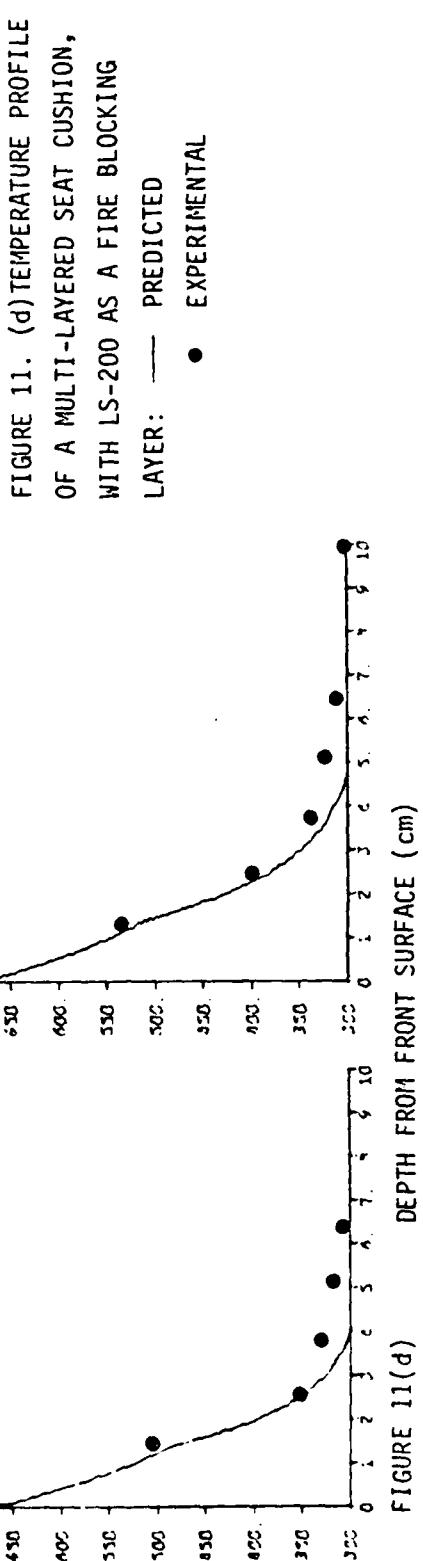
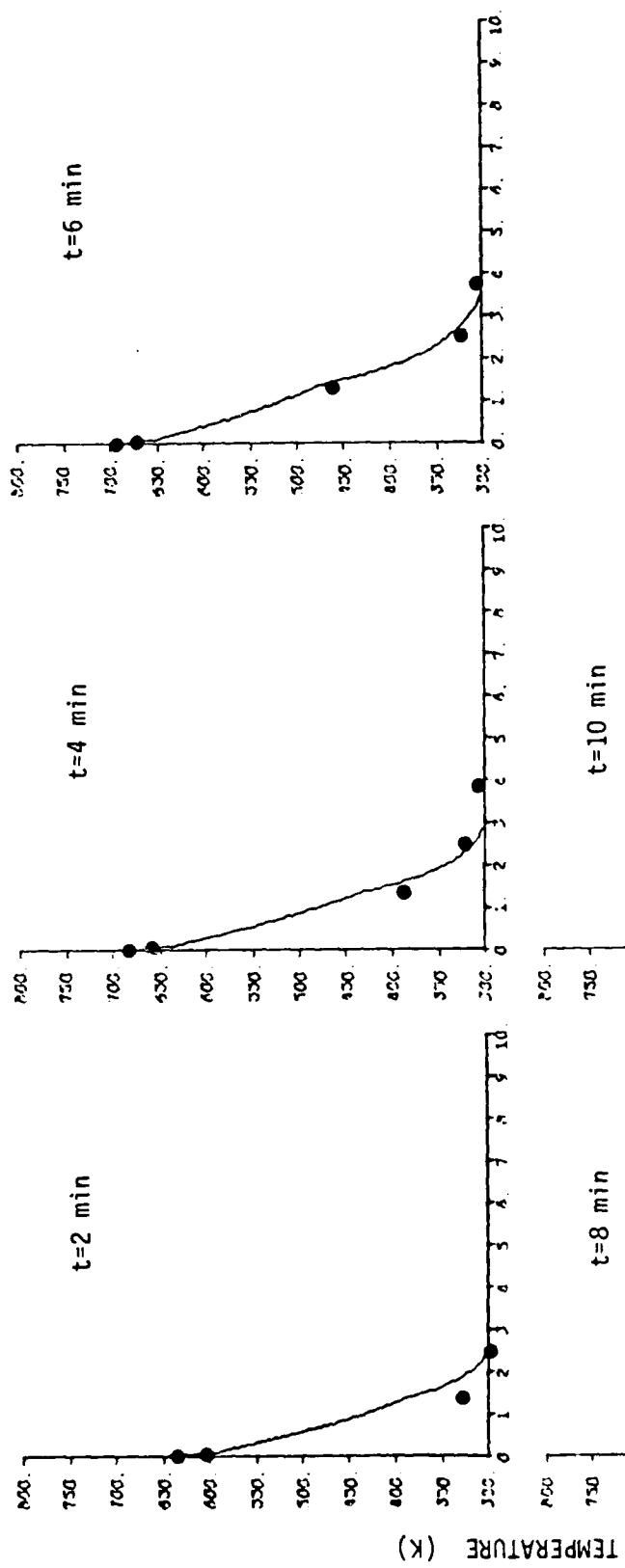


FIGURE 11(d) TEMPERATURE PROFILE OF A MULTI-LAYERED SEAT CUSHION, WITH LS-200 AS A FIRE BLOCKING LAYER: — PREDICTED ● EXPERIMENTAL

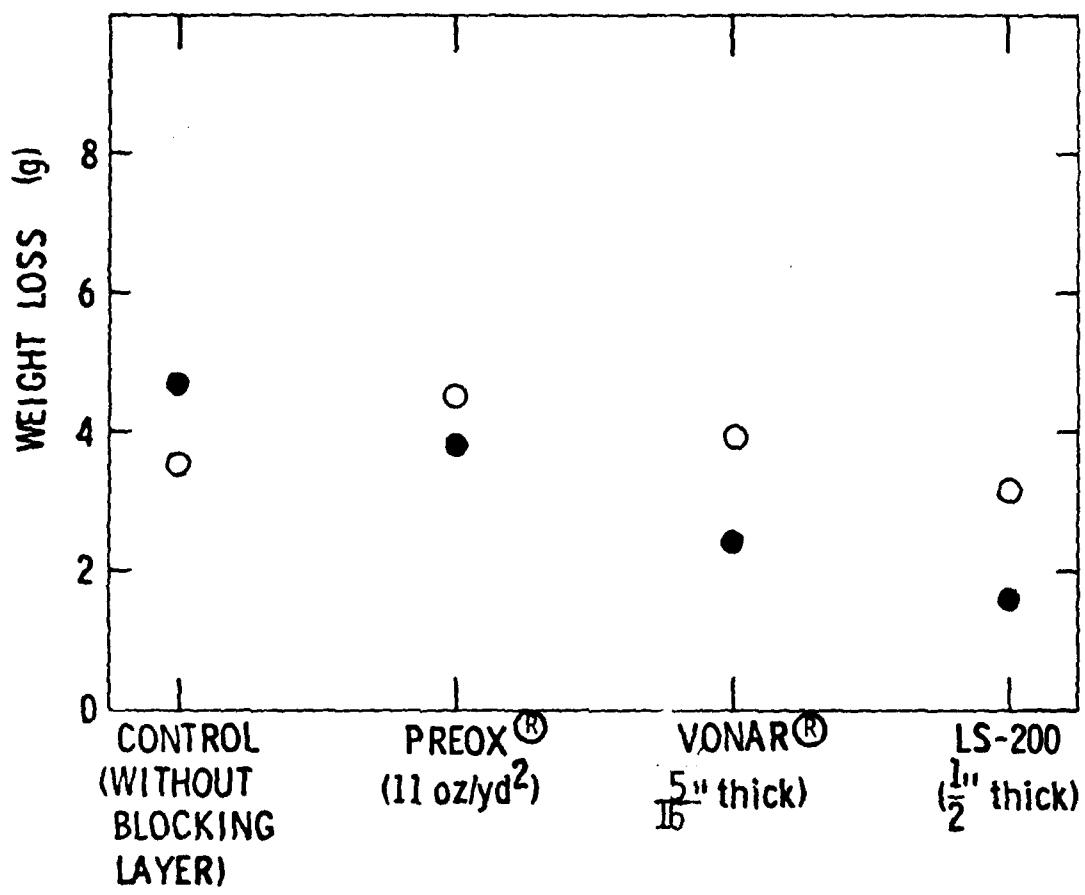
where the depth is large. The positive deviations, i.e., temperatures measured higher than predicted, are seen wherever the void is observed inside of the polyurethane foam. This is again believed to be caused by a convective flow of hot combustion gas within the void, which leads to increased heat transfer to the colder side.

The weight losses calculated by the model are compared with the experimental data, as shown in figure 12. The figure shows that the calculated weight loss is about 15 to 95 percent higher than the measured values in all cases, but the order of the mass loss is in agreement with the experimental data except for the control (without a fire blocking layer). This again is due to the presence of the void formed inside of the foam, which is clearly the physical state this thermochemical model is not aimed to be used for.

## 5. SUMMARY

The work on the prediction of thermochemical performance of multi-layered seat cushion materials leads to four conclusions:

1. The concept of thermochemical sublayers is applicable to describe the complex pyrolysis behavior observed for many of the currently used and proposed aircraft interior polymeric materials.
2. The thermochemical model of the past can be extended to handle such multilayered systems analytically.
3. The model predicts reasonably within a factor of 2 both the weight loss due to burning and the temperature profile established within the seat materials.
4. The thermochemical model, can be used with the minimum number of input data determined by experiments for the thermochemical performance prediction of other multilayered materials under fire conditions. This analysis coupled with experiments in the NBS Smoke Chamber offers a useful small-scale test procedure for evaluating candidate blocking layer materials.



## REFERENCES

1. Howell, W., Opening Remarks in Workshop on Mathematical Fire Modeling, March 24-27, 1981, proceedings, Aug. 1981, U. S. Department of Transportation, Federal Aviation Administration, Report No. DOT/FAA/CT-81/209.
2. Sarkos, C., Aircraft Fire Scenarios, ibid.
3. Eklund, T., FAA Modeling Efforts, ibid.
4. Dokko, W. and Ramohalli, K., Application of Thermochemical Modeling to Aircraft Interior Polymeric Materials, U.S. Department of Transportation, Final Report No. DOT/FAA/CT-82/83.
5. Ramohalli, K. and Mink, M., Thermal Performance Modification of Composite Materials, AIAA 17th Aerospace Sciences Meeting, New Orleans, Louisiana, January 1979, Paper No. 79-0018.
6. Kumar, R. N. and Stickler, D. B., Polymer Degradation Theory of Pressure- Sensitive Hybrid Combustion, XIII Symposium (International) on Combustion, The Combustion Institute, 1971, pp. 1059-1072.
7. Kourtides, D. A. and Parker, J. A., Test Methodology for Evaluation of Fireworthy Aircraft Seat Cushions, J. Fire & Flamm., Vol. 15, pp. 56-76, 1982.
8. Woolridge, C.E. and Muzzy, R.J., "Experiments in a Turbulent Boundary-Layer with transpiration on Combustion," X Symposium (International) in Combustion, The Combustion Institute, 1965, p. 1337.
9. Kulgein, p. 417, JFM 12 (1962).
10. Fennimore, C.P. and Jones, G.W., C&F, X, p. 295 (1966).
11. Spalding, D.B., "Some Fundamentals of Combustion," Butterworth, London, (1954).

## APPENDIX A Derivation of $r$

The differential equations are

$$k \frac{d^2T}{dx^2} + \rho_{cr} \frac{dT}{dx} = D\rho_{NB} \exp(-E/RT) \quad (A-1)$$

$$- \frac{1}{N} \frac{dN}{dt} = B \exp(-E/RT) \quad (A-2)$$

and the boundary conditions are

$$\text{at } x = 0 \quad T = T_s \quad (A-3)$$

$$N = N_s = 1 - \frac{1}{FS_s}$$

$$\text{at } x = x_L \quad T = T_c \quad (A-4)$$

$$N = N_c = 1 - \frac{1}{FS_c}$$

By introducing the dimensionless terms,

$$y = \frac{\rho_{cr} x}{k} \quad (A-5)$$

$$\text{and } \tau = \frac{T - T_0}{T_s - T_0} \quad (A-6)$$

Then Eqs. (A-1) and (A-2) become

$$\frac{d^2\tau}{dy^2} + \frac{d\tau}{dy} = \frac{k D \rho_{NB} \exp(-E/RT)}{(\rho_{cr})^2 (T_s - T_0)} \quad (A-7)$$

$$\frac{dN}{d\tau} \cdot \frac{d\tau}{dy} = \frac{k c \rho_{NB} \exp(-E/RT)}{(\rho_{cr})^2} \quad (A-8)$$

Further use of dimensionless variables,

$$p = \frac{d\tau}{dy} \quad (A-9)$$

$$h = \frac{D}{c(T_s - T_0)} \quad (A-10)$$

$$\text{and } \Lambda = \frac{kB}{\rho cr^2} \quad (A-11)$$

convert the Eqs. (A-7) and (A-8) to

$$pp' + p = \Lambda h N \exp(-E/RT) \quad (A-12)$$

$$pN' = \Lambda N \exp(-E/RT) \quad (A-13)$$

where a prime denotes  $d/d\tau$ .

The boundary conditions are accordingly

$$\tau = 1, \quad p = p_s, \quad N = N_s \quad (A-14)$$

$$\tau = \tau_c, \quad p = p_c, \quad N = N_c \quad (A-15)$$

The Eq. (A-13) can be combined to Eq. (A-12) to yield

$$pp' + p - pN'h = 0 \quad (A-16)$$

Assuming  $p \neq 0$ , then Eq. (A-16) becomes essentially

$$\frac{dp}{d\tau} + 1 - \frac{dN}{d\tau} = 0 \quad (A-17)$$

If Eq. (A-17) is integrated using the boundary conditions of Eqs. (A-14) and (A-15), then

$$N = (p + \tau - p_s - 1 + h N_s)/h \quad (A-18)$$

$$\text{or } N = (p + \tau - p_c - \tau_c + h N_c)/h \quad (A-19)$$

Then substituting Eq. (A-19) into Eq. (A-12) gives

$$pp' + p = \Lambda(p + \tau - p_s - 1 + h N_s) \exp(-E/RT) \quad (A-20)$$

This is the differential equation to be solved with the boundary equations (A-14) and (A-15).

If the  $T$  is close to  $T_s$ , then as an approximation, Eq. (A-20) can be written as

$$pp' + p = (p + \tau - p_s - 1 + h N_s) \Lambda \exp \{ -\Theta [1 + \chi (1-\tau)] \} \quad (A-21)$$

where  $\Theta = \frac{E}{RT_s}$ , (A-22)

and  $\chi = \frac{T_s - T_0}{T_s}$  (A-23)

Let  $\lambda = \Lambda \exp(-\Theta)$  (A-24)

and  $g = -p_s - 1 + h N_s$  (A-25)

then, the Eq. (A-21) is

$$pp' + p = (p + \tau + g) \lambda \exp[-\Theta \chi (1-\tau)] \quad (A-26)$$

Further defining

$$\eta = \frac{1-\tau}{\epsilon} \quad (A-27)$$

$$\xi = p + \tau \quad (A-28)$$

$$\lambda = \epsilon \lambda \quad (A-29)$$

where  $\epsilon = \frac{1}{\Theta \chi}$  (A-30)

enables rewriting Eq. (A-26) as

$$- (\xi - 1 + \epsilon\eta) \xi' = (\xi + g) \lambda \exp(-\eta) \quad (A-31)$$

or

$$\frac{d\xi}{\xi + g} - \frac{\xi d\xi}{\xi + g} = \lambda e^{-\eta} d\eta \quad (A-32)$$

The boundary conditions for this equation are

$$\eta = 0 \quad \xi = \xi_s = p_s + 1 \quad (A-33)$$

$$\eta = \eta_c \quad \xi = \xi_c = p_c + T_c \quad (A-34)$$

Integrating now with respect to  $\xi$  and  $\eta$ , in the region between two surfaces,  $s$  and  $c$ , is

$$\int_c^s \frac{d\xi}{\xi + g} - \int_c^s \left( \frac{\xi}{\xi + g} \right) d\xi = \int_c^s \lambda e^{-\eta} d\eta \quad (A-35)$$

to give the result as

$$\ln \left( \frac{\xi_s + g}{\xi_c + g} \right) - \left\{ (\xi_s - \xi_c) - g \ln \left( \frac{\xi_s + g}{\xi_c + g} \right) \right\} = -\lambda (1 - e^{-\eta_c}) \quad (A-36)$$

Then  $\lambda$  is simply given by

$$\lambda = \frac{1}{(1 - e^{-\eta_c})} \cdot \left\{ (\xi_s - \xi_c) + (1 + g) \ln \left( \frac{\xi_c + g}{\xi_s + g} \right) \right\} \quad (A-37)$$

By combining Eq. (A-29), (A-30), (A-22), (A-23) and (A-11),  $r$  can be shown in more familiar terms,

$$r = \sqrt{\frac{(k/p_c) B \exp(-E/RT_s)}{\lambda \left( \frac{E}{RT_s} \right) \left( \frac{T_s - T_c}{T_s} \right)}} \quad (A-38)$$

## APPENDIX B Computer Program

```

1.      C      PROGRAM A
2.      C      THIS PROGRAM TAKES THE INTERFACIAL TEMPERATURES (B.C.'S) AS THE INPUT
3.      C      DATA AND THEN CALCULATES THE BURNING RATE, HEAT FLUX, TEMPERATURE PROFILE
4.      C      AND WEIGHT LOSS OF MULTI-LAYERED POLYMERIC MATERIALS (INCLUDING
5.      C      THE FIRE BLOCKING LAYER) USED FOR AIRCRAFT SEAT CUSHIONS.
6.      C      THE DIMENSIONS USED IN THIS PROGRAM ARE GRAM, GRAM-MOLE, 0-CELSIUS,
7.      C      CALORIES, CM, AND SECOND.
8.      C
9.      C      DIMENSION ANAME(7,2),Y(7),TFF(7),KT(7,11),CPT(7,11),
10.     C      - RHOT(7,11),HT(7,11),ET(7,11),DT(7,11),HEATRT(7,11),
11.     C      - TKL(7,11),TKH(7,11),TL(7,11),TH(7,11),
12.     C      - FL(7,11),FKH(7,11),FL(7,11),FH(7,11),
13.     C      - NKLO(7,11),NKHI(7,11),NL(7,11),NH(7,11),
14.     C      - JSTAFT(11),JEND(11),HEADIN(14),ID(11),
15.     C      - NS(11),NC(11),ANAME(11,2),T(11),K(11),CP(11),RH0(11),R(11),
16.     C      - E(11),D(11),HEATRA(11),FSS(11),FSC(11),RE(11),CIN(11),OCUT(11),
17.     C      - QNET(11),TP(11,11),NSSP(11,11),TSTART(11),TINTV(11)
18.     C      DIMENSION TPLOT(121),PLGT(121),RHS(11,11),DXDT(11,11),DY(11,11)
19.     C      DIMENSION X(11,11)
20.     C      DIMENSION H(11),PS(11),PC(11),YIS(11),YIC(11),THETA(11),ETAC(11)
21.     C      DIMENSION ENC(11),LAM5UA(11),CH(11),ALPHMA(11)
22.     C      DIMENSION TDC(11),RDC(11),A(11),G(11),RBT(11)
23.     C      REAL KT,K,MW,VS,NC,NKLO,NKHI,NL,NH,LAMBDA
24.     C
25.     C      READ IN THE INGREDIENT MATERIAL PROPERTY AND PRINT OUT
26.     C
27.     C      READ (5,901) ITOTAL
28.     901 FORMAT (12)
29.     DO 100 11=1,ITOTAL
30.       READ (5,902) I,ANAME(1,1),ANAME(1,2),MW(1),NTREG(1)
31.     902 FORMAT(12.2E5,F10.2,12)
32.       NTR=NTREG(1)
33.       READ (5,903)(TKL(I,J),TKH(I,J),KT(I,J),CPT(I,J),RHOT(I,J),
34.       - BT(I,J),ET(I,J),DT(I,J),HEATRT(I,J),J=1,NTR)
35.     903 FORMAT (2F5.1, 7E10.3)
36.     100 CONTINUE
37.     WRITE (6,948)
38.   948 FORMAT (1H1, 25Y, *BURNING RATE PREDICTION FOR MULTI-LAYERED*

```

```

*   * AIRCRAFT SEAT CUSHION* / 26X, E3(*=*), /
*   5X, *MATERIAL PROPERTY* / 21X, *TEMP. RANGE, THERMAL*, 5X,
*   *HEAT*, 6X, *DENSITY, FREEEXP, ACTIVATION HEAT OF HEAT*, 5X,
*   / 36Y, *CONDUCTIV. CAPACITY*, 13X, *FACTOR ENERGY*, 5X,
*   7X, *DECOMPOS. RATE*, 5X, *K*(*CAL/CM.S.C) (*CAL/GM.C) (GM/CM3)*, 13Y,
*   -26X, *(K)*, 5X, *(CAL/CM.S.C) (*CAL/GM.C) (GM/CM3)*, 13Y,
*   -*(CAL/MOLE) (*CAL/GM) (DEG C/SFC)* /
*   DO 102 I=1,ITOTAL
*   NTR=NTREC(I)
*   WRITE (6,951) I,ANAMET(I,1),ANAMET(I,2)
*   951 FORMAT (5X,I2,2),2A5)
*   WRITE (6,953) (TKLO(I,J),TKHI(I,J),
*   - KT(I,J),CFT(I,J),RHT(I,J),PT(I,J),ET(I,J),DT(I,J),HEATPT(I,J),
*   - J=1,NTR)
*   953 FORMAT (17X, F7.1, *-* , F7.1, 1Y,
*   - F10.5,3Y, F6.3, 3Y, F6.3, 2Y, E11.3, 4X, F9.0, F10.1, F10.2)
*   1C2 CONTINUE
*   C
*   C FIX CONSTANTS USED IN THIS PROGRAM
*   C
*   P=1.0
*   P=1.9P717
*   PM=82.06
*   CGAS=0.382
*   EPSLON=0.9
*   SIGMA=1.355E-12
*   T0=225.0
*   C
*   C CALCULATE N AND FS AT TWO END TEMPERATURES OF KINETIC REGIMES
*   C THE CALCULATION SEQUENCE IS, BY LOGIC, FROM LOW TO HIGH TEMPERATURE
*   C
*   WRITE (6,955)
*   955 FORMAT (1W1, 10X, *NORMALIZED FFAGMENT SIZE AT END TEMPS OF*, 1Y,
*   - *KINETIC REGIMES*, /, 11X, 56(*=*), /)
*   DO 200 I=1,ITOTAL
*   WRITE (6,951) I, ANAMET(I, 1), ANAMET(I,2)
*   NTR=NTREC(I)
*   NKLO(I,NTR)=1.0
*   FKLO(I,NTR)=10000.
*   DO 205 J=1,NTR
*   J=NTR+1-JJ
*   CALL FFAGPT (RT(I,J),ET(I,J),TKLO(I,J),TKHI(I,J),
*   - HEATPT(I,J),NKLO(I,J),NKHI(I,J))
*   IF (J,NE.1) NKLO(I,J-1)=NKHI(I,J)
*   IF (NKLO(I,J).EQ.1.0) NKLO(I,J)=0.9999599
*   FKLO(I,J) = 1.0/41.0-NKLO(I,J)
*   IF (NKHI(I,J).EQ.1.0) NKHI(I,J)=0.9999599
*   FKHI(I,J) = 1.0/(1.0-NKHI(I,J))
*   WRITE (6,952) TKLO(I,J),TKHI(I,J),NKLO(I,J),NKHI(I,J)
*   958 FORMAT (22X, F7.1, *-* , F7.1, 5X, F11.6, 2X, *--*, 1X, F11.6)
*   205 CONTINUE
*   200 CONTINUE
*   C
*   C INITIALIZE A FEW VARIABLES FOR COMPUTER PLOTTING
*   C
*   NPAGE=0
*   NPLOTE=1

```

```

5*      IPLOT=C
6*      CALL RGNPLT
7*      C
8*      C  READ IN TEMPERATURE DATA, I.E.,
9*      C  TFMP. OF TOP SURFACE OF EACH LAYER AND REAR SURFACE (COLD)
10*      C
11*      1111 CONTINUE
12*      WRITE (6,898)
13*      898 FORMAT (1H1, 25X, *URNING RATE PREDICTION FOR MULTI-LAYERED*
14*      - * AIRCRAFT SEAT CUSHION* / 26X, E3(0=0), /)
15*      READ (5,904,END=1112) TBACK,(HEADIN(I),I=1,14)
16*      904 FORMAT (F10.0,14A5)
17*      WRITE (6,899) (HEADIN(I),I=1,14)
18*      899 FORMAT (40X, 14A5, //)
19*      -      5), *MATERIAL*      7X, *SURFACE      THERMAL      HEAT
20*      -      DENSITY      PREEXP.      ACTIVATION      HEAT OF      HEAT
21*      -      21X, *TEMP.      CONDUCTIV.      CAPACITY      FACTOR      ENERGY
22*      -      DECOMPOS.      RATE*/*
23*      -      21X, *(K)*, 5X,* (CAL/CM.S.C)  (CAL/GM.C)  (G4/CH3)*, 13X,
24*      -      *(CAL/MOLE)  (CAL/CM)  (DEG C/SEC)* //)
25*      I=1
26*      105 CONTINUE
27*      READ (5,906,END=110) ID(I),TH(I)
28*      906 FORMAT (12.F10.2)
29*      I=1+1
30*      GO TO 105
31*      110 CONTINUE
32*      NLAYG=I-1
33*      C
34*      NLAYG1=NLAYG-1
35*      DO 115 I=1,NLAYG1
36*      115 TLO(I)=TH(I+1)
37*      TLO(NLAYG)=TBACK
38*      C
39*      C  NOW CALCULATE THE FRAGMENT SIZES AT THE TWO BOUNDARIES OF EACH
40*      C  GENERIC LAYER
41*      C
42*      DO 300 I=1,NLAYG
43*      IDENT=ID(I)
44*      NTID=NTREC(IDENT)
45*      IF (TLO(I).GE.TKL0(IDENT,NTID)) GO TO 301
46*      JSTART(I)=NTID
47*      KLO(I)=1.0
48*      FLO(I)=10000.
49*      GO TO 304
50*      301 CONTINUE
51*          DO 302 J=1,NTID
52*          IF (TLO(I).GE.TKL0(IDENT,J).AND.TLO(I).LT.TKH0(IDENT,J))
53*          GO TO 305
54*      302 CONTINUE
55*      WRITE (6,961)
56*      961 FORMAT (5X,*FOR *, 12, *TH LAYER, TLO IS OUT OF BOUND*)
57*      GO TO 304
58*      305 JSTART(I)=J
59*      CALL FPAGHT(PT(IDENT,J),ET(IDENT,J),TKL0(IDENT,J),TLO(I),
60*      -      HEATRT(IDENT,J),NKLO(IDENT,J),NL0(I))
61*      FLO(I) = 1.0/(1.0-VLO(I))
62*

```

```

130 304 CONTINUE
140 IF (THI(I).GE.TKLOC(IDENT,NTID)) GO TO 306
150 JEND(I)=NTID
160 NH1(I)=1.0
170 FHI(I)=10000.
180 GO TO 300
190 306 CONTINUE
200 DO 307 J=1,NTID
210 IF (THI(I).GE.TKLOC(IDENT,J).AND.THI(I).LT.TKHI(IDENT,J))
220 GO TO 308
230 307 CONTINUE
240 WRITE (6,962) I
250 962 FORMAT (5X, *FOR *, I2, *TH LAYER, THI IS OUT OF BOUND*)
260 308 JFND(I)=J
270 CALL FRAGHT(THI(IDENT,J),ETI(IDENT,J),TKLOC(IDENT,J),THI(I),
280 - HEATRT(IDENT,J),TKLOC(IDENT,J),NH1(I))
290 FHI(I)=1.0/(1.0-NH1(I))
300 CONTINUE
310 C
320 C REDEFINE AND RENUMBER THE SUBDIVIDED MULTILAYERS, BASED ON THE
330 C TEMPERATURE RANGES OF THERMOCHEMICAL REGIMES AND ALSO ON THE GENERIC
340 C DIVISIONS
350 C SEQUENCE OF IJ IS FROM FRONT (HOTTEST) LAYER TO REAR (COLDEST) LAYER
360 C
370 IJ=0
380 DO 310 I=1,NLAYG
390 IDENT=ID(I)
400 JENDI=JEND(I)
410 JSTA1=JSTART(I)
420 IJ=IJ+1
430 T(IJ)=THI(I)
440 ANAMF(IJ,1)=ANAMET(IDENT,1)
450 ANAME(IJ,2)=ANAMET(IDENT,2)
460 D(IJ)=DT(IDENT,JENDI)
470 CP(IJ)=CPT(IDENT,JENDI)
480 RHO(IJ)=RHOT(IDENT,JENDI)
490 E(IJ)=ET(IDENT,JENDI)
500 B(IJ)=BT(IDENT,JENDI)
510 K(IJ)=KT(IDENT,JENDI)
520 HEATRA(IJ)=HEATRT(IDENT,JENDI)
530 AS(IJ)=NH1(I)
540 FSS(IJ)=FHI(I)
550 NC(IJ)=RKLC(IDENT,JENDI)
560 FSC(IJ)=FKLC(IDENT,JENDI)
570 IF (JENDI.EQ.JSTA1) NC(IJ)=AL0(I)
580 IF (JENDI.EQ.JSTA1) FSC(IJ)=FL0(I)
590 IF (JENDI.EQ.JSTA1) GO TO 310
600 JENDI1=JENDI+1
610 DO 311 J=JENDI1,JSTA1
620 IJ=IJ+1
630 T(IJ)=TKHI(IDENT,J)
640 ANAMF(IJ,1)=ANAMET(IDENT,1)
650 ANAME(IJ,2)=ANAMET(IDENT,2)
660 D(IJ)=DT(IDENT,J)
670 CP(IJ)=CPT(IDENT,J)
680 RHO(IJ)=RHOT(IDENT,J)
690 E(IJ)=ET(IDENT,J)

```

```

10*          B(IJ)=FT(IDENT,J)
11*          K(IJ)=KT(IDENT,J)
12*          HEATRA(IJ)=HEATRI(IDENT,J)
13*          NS(IJ)=NKHI(IDENT,J)
14*          FSS(IJ)=FKHI(IDENT,J)
15*          NC(IJ)=LKLO(IDENT,J)
16*          FSC(IJ)=FKLO(IDENT,J)
17*          IF (J.GE.1,USTAI)  NC(IJ)=NL0(I)
18*          IF (J.GE.1,USTAI)  FSC(IJ)=FL0(I)
19*          311  CONTINUE
20* 310  CONTINUE
21*          NLAYER=IJ
22*          T(IJ+1)=TBACK
23*          LPITE (6,500) (I,4NAME(I,1),A,NAME(I,2), T(I), K(I), CP(I), RHO(I),
24*          - B(I), E(I), D(I), HEATRA(I), I=1,NLAYER)
25* 500  FORMAT (EX, 12.2X, 24S, F6.1, 1Y, F10.5, 4Y, F6.3, 5X, F6.3,
26*          - E14.3, 3Y, F6.0, 4X,F7.0,F9.3 )
27*          WRITE (6,907) TBACK
28* 907  FORMAT (*), *KEAP SURF*, F6.0, // )
29*          WRITE (6, 910)
30* 910  FORMAT (5/, *PREDICTED BURNING RATE* //,20X, *BURN RATE*
31*          - * SURFACE TEMP. HEAT*
32*          - * FLUX IN  HEAT FLUX OUT  HEAT CONSUMED  FS-TOP SURFACE*
33*          - * FS=HGTDM / 21.1, *(CM/SEC)*, 7X, *(K)*, 7X,
34*          - 31 *(CAL/CM2SEC)*, 3Y 3, // )
35*          NLAY1 = NLAYER - 1
36*          FCALC = 0
37*          C
38*          C      CALCULATE THE BURNING RATES FOR THE FIRST TIME ASSUMING THAT THE BURN
39*          C      RATE ARE THE SAME FOR EVERY LAYER
40*          C
41*          DO 333 I=1, NLAYER
42*          T10(I) = T(I) - T0
43*          H(I) = D(I) / (CP(I) + (T(I)-T0))
44*          ALPHA(I) = K(I) / (RHO(I)*CP(I) )
45*          CH(I) = (T(I)-T0) / T(I)
46*          THETA(I) = E(I) / (R*T(I))
47*          XIS(I) = -H(I) / FSS(I)
48*          XIC(I) = -H(I) / FSC(I)
49*          TAUC = (T(I+1) - T0) / (T(I) - T0)
50*          IF (T(I+1).LT.0.0) GO TO 337
51*          ETAC(I) = THETA(I) + CH(I) + (1.0 - TAUC)
52*          ENC(I) = EXP(-ETAC(I))
53*          HYICS=(H(I)*YIC(I)) / (H(I) + YIS(I) )
54*          LAMBDA(I) = (XIS(I)-YIC(I)) + (1.0+H(I))* ALOG(HYICS)/(1.0-ENC(I))
55*          PSQ = K(I) + E(I) + EXP(-THETA(I)) / (RHO(I)*CP(I)* LAMBDA(I) *
56*          - THETA(I) + CH(I))
57*          IF (RSQ.LT.0.0) WRITE(6,950) I,RSQ,LAMBDA(I)
58*          IF (RSQ.LT.0.0) RSQ=-RSQ
59*          GO TO 500
60* 337  CONTINUE
61*          RSQ = ALPHA(I)*B(I)*EXP(-THETA(I))/(THETA(I)*CH(I)*((1.0+H(I))*
62*          - ALOG(FSS(I)/(FSS(I)-1.0)) + XIS(I)))
63*          IF (RSQ.LT.0.0) WRITE (6,950) I,RSQ,FSS(I)
64*          IF (RSQ.LT.0.0) RSQ=-RSQ
65* 500  CONTINUE
66*          RR(I) = SQRT(RSQ)

```

```

67*      PS(I) = -1.0 + XIS(I)
68*      GIN(I) =-FS(I) + T10(I) + PR(I)*RHO(I)*CP(I)
69*      FC(I) = -TAUC + YIC(I)
70*      QDUT(I) =-FC(I)* T10(I) + RF(I)*RHO(I)*CP(I)
71*      QNET(I) = QIN(I) - QDUT(I)
72*      333 CONTINUE
73*      C      CALCULATE PUPMING RATES SIMULTANEOUSLY FOR ALL THE LAYERS
74*      C
75*      379 CONTINUE
76*      IF (R.CALC.EQ.100) GO TO 777
77*      DO 381 I=1,NLAYER
78*          RBT(I) = RR(I)
79*          RRC(I) = RPT(I) + RHC(I) + CP(I)
80*          A(I) = RHC(I)*(T(I)-T(I+1)) + D(I)*RHO(I)*RBT(I)
81*          - + (1./FSS(I) - 1./FSC(I))
82*      381 CONTINUE
83*      WRITE (6,524)
84*      C 524 FORMAT (1/4, "I", E, "FS", 11, "XIS", 11, "PC", 11, "YIC", 11, "G",
85*      C - 11X, "ENC", 11, "CR", 11, "LAMRDA")
86*      DO 382 II=1,NLAY1
87*          I = NLAY1 + 1 - II
88*          FS(I) = 0.0
89*          DO 384 J=1, NLAYER
90*              FS(I) = PS(I) + A(J)
91*              PS(I) = -PS(I) / (T10(I)*RHOCP(I))
92*              G(I) = - (FS(I) + 1.0 - H(I)*(1.0 - 1.0/FSS(I)) )
93*              XIS(I) = PS(I) + 1.0
94*              FC(I) = PS(I) + A(I)/ (T10(I)*RRC(I) )
95*              YIC(I) = PC(I) + (T(I+1)-T(I))/T10(I)
96*              ETAC(I) = (E(I)/R)*(T(I)-T(I+1))/(T(I)*T(I))
97*              ENC(I) = EXP(-ETAC(I))
98*              GXICS = (G(I)*YIC(I)) / (G(I) + XIS(I) )
99*              IF (GXICS.LT.0.0) WRITE (6,918) I,GXICS,G(I),YIC(I),XIS(I)
100*             IF (GYICS.LT.0.0) GO TO 777
101*             918 FORMAT (10X,"GXICS", "12,0", "F15.6, "15.6, "YIS", "15.6)
102*             LAMRDA(I) = (1./IS(I)*YIC(I) + (1.0+G(I))*ALOG(GXICS))/ (1.0-ENC(I))
103*             RSQ = ALPHA(I)*H(I)*EXP(-THETA(I))/(LAMRDA(I)*THETA(I)*CH(I))
104*             IF (RSQ.LT.0.0) PS0=-RSQ
105*             RB(I) = SQRT(RSQ)
106*             RHC(I) = RB(I) + RHO(I) + CP(I)
107*             A(I) = RHC(I)*(T(I)-T(I+1)) + D(I)*RHO(I)*RR(I)
108*             - + (1./FSS(I) - 1./FSC(I))
109*             C      WRITE (6,925) I,PS(I),XIS(I),PC(I),YIC(I),G(I),ENC(I),
110*             C - LAMRDA(I)
111*             C 925 FORMAT (3X, I2, 8(3X,E10.5) )
112*             C      GIN(I) =- PS(I) + (T(I)-T(I+1))*RHC(I)*RHO(I)*CP(I)
113*             C      QDUT(I) =-PC(I)*(T(I)-T(I+1)) + RP(I)*RHO(I)*CP(I)
114*             C      QNET(I) = QIN(I) - QDUT(I)
115*             382 CONTINUE
116*             WRITE (6,920) I, ANAME(I,1), ANAME(I,2), RB(I), T(I),
117*             - QIN(I), QDUT(I), QNET(I), FSS(I), FSC(I), I=1,NLAYER)
118*             C
119*             C      STOP IF CONVERGED ENOUGH OR KEEP ITERATIVE CALCULATION
120*             C
121*             DO 386 I=1,NLAYER
122*             IF (APS(RRT(I)/RP(I)-1.0).LT.0.01) GO TO 386

```

```

24*      NCALC = NCALC + 1
25*      GO TO 379
26* 386 CONTINUE
27* 777 CONTINUE
28*      WRITE (6,928) NCALC
29* 928 FORMAT (2Y, *% OF ITERATION = * 130 /)
30*      DO 388 I=1,NLAYER
31*      QIN(I) =- FS(I) * (T(I)-TC)*RP(I)*RH0(I)*C(I)
32*      QOUT(I) =- PC(I)* (T(I)-TC) * RE(I) * RH0(I) * .PEI
33*      QNET(I) = QIN(I) - QOUT(I)
34* 388 CONTINUE
35*      WRITE (6,920) (I, ANAME(I,1), ANAME(I,2), F(1), T(1),
36*      - QIN(I), QOUT(I), QNET(I), FSS(I), FSC(I), I=1,NLAYER)
37* 920 FORMAT (EX,12.2Y,245,F10.5, 2X, F10.1, 3Y, F10.5, 2X, F10.4, 1C,
38*      - F10.4, 4Y, F10.5, 4Y, F10.5 /)
39* 920 FORMAT (10X,*FSG (*,12, 0)= *, F13.5, 10Y, 12, 0)
40*      WLLOSS=0.0
41*      DO 393 I=1,NLAYER
42* 393 WLLOSS=WLLOSS+PC(I)*RH0(I)*(1./FSS(I)-1./FSC(I))
43*      WRITE (6,927) WLLOSS
44* 927 FORMAT (//, * WEIGHT LOSS PREDICTED BY THE MODEL * /3 *741*2)
45*      - * /, 4D9,F10.5, * GRAM/SEC.CM2 OF BURNING SURFACE * )
46*      C
47*      C PRELIMINARY CALCULATION BEFORE PLOTTING T VS X
48*      C
49*      DO 1120 I=1,NLAYER
50*      TINTV(I)=(T(I)-T(I+1))/FLOAT(10)
51*      TP(I,1)=T(I)
52*      DO 1121 J=1+10
53*      TP(I,J+1)=T(I)-TINTV(I)*FLOAT(J)
54* 1121 CONTINUE
55* 1120 CONTINUE
56*      DO 1210 I=1,NLAYER
57*      NSSP(I,1)=NC(I)
58*      DO 1220 JJ=2+11
59*      J=12-JJ
60*      IF(TP(I,J).LE.TSTART(I)) NSSP(I,J)=1.0
61*      IF(TP(I,J).LE.TSTART(I)) GO TO 1220
62*      CALL FRAGMT((I),E(I),TP(I,11),TP(I,J),HEATR(I),NSSP(I,11),
63*      - NSSP(I,J))
64* 1220 CONTINUE
65* 1210 CONTINUE
66*      RHS(NLAYER+11)=0.00001
67*      DO 1310 II=1,NLAYER
68*      I=NLAYER+1-II
69*      RHORB=RH0(I)*RB(I)
70*      DO 1320 JJ=2+11
71*      J=12-JJ
72*      DTDXK=RHORB*((CF(I)*(TP(I,J)-TP(I,J+1))
73*      - D(I)*(NSSP(I,J+1)-NSSP(I,J)))
74*      RHS(I,J)=DTDXK+FHS(I,J+1)
75*      DXNT(I,J)=K(I)/RHS(I,J)
76* 1320 CONTINUE
77*      RHS(I-1+11)=FHS(I,J)
78* 1310 CONTINUE
79*      SUM=0.0
80*      DO 1330 II=1,NLAYER

```

```

91*      I=NAYER,P+1-I]
92*      DO 1330  JJ=1,5,2
93*      J=12-JJ
94*      DX(I,J-2)=(DXDT(I,J)+4.*DXDT(I,J-1)+DXDT(I,J-2))+TINTV(I)/3.
95*      SUM=SUM+DX(I,J-2)
96*      X(I,J-2)=SUM
97*      1330  CONTINUE
98*      C  CLEAR THE MEMORIES OF XPLOT AND TPLOT
99*      DO 1336  I=1,121
100*      XPLOT(I)=0.0
101*      1338  TPLOT(I)=0.0
102*      DO 1340  I=1,ALAYER
103*      DO 1340  JJ=1,5
104*      IIND=(I-1)*5 + JJ
105*      J=JJ*2-1
106*      XPLOT(IIND)=SUM - X(I,J)
107*      TPLOT(IIND)=TP(I,J)
108*      1340  CONTINUE
109*      IIND=IIND+1
110*      XPLOT(IIND)=SUM
111*      TPLOT(IIND)=TP(I,J+2)
112*      IPLOT=IPLOT+1
113*      IF (IPLOT.GT.5) IPLOT=1
114*      IF (IPLOT.EQ.1.AND.NPLOT.NE.NPAGE) CALL ADVPLT
115*      CALL PLFORM ('LINLIN',2.5,2.75)
116*      CALL PLABEL ('TEMPERATURE PROFILE', 0,
117*      - 'DEPTH FROM SURFACE (CM)', 0, 'TEMPERATURE (K)', 0)
118*      CALL PLSCAL ((0.0,10.0)*2,100000,(300.0,800.0),2,100000)
119*      CALL PLCSIZ(0.06,0.0)
120*      GO TO (1341,1342,1343,1344,1345), IPLOT
121*      1341  CALL ORIGIN (1.0,4.3)
122*      GO TO 134P
123*      1342  CALL ORIGIN (3.0,0.0)
124*      GO TO 134E
125*      1343  CALL ORIGIN (3.0,0.0)
126*      GO TO 134E
127*      1344  CALL ORIGIN (-6.0,-3.1)
128*      GO TO 134P
129*      1345  CALL ORIGIN (3.0,0.0)
130*      NPLOT=NPLOT+1
131*      134E  CALL PLGRAF
132*      CALL PLCURV (XPLOT,TPLOT,121,0,***)
133*      GO TO 1111
134*      1112  CONTINUE
135*      CALL ENDPLT
136*      STOP
137*      END

```

```
1*      SUBROUTINE FRAGMT(SH,SE,STI,STF,SHR,SNI,SNF)
2*      DOUBLE PRECISION SY1,SX2,SY1,SY2,DEI
3*      STHI=SE/(1.98717*STI)
4*      STHF=SE/(1.98717*STF)
5*      SA1=EXP(-STHF)*STF - EXP(-STHI)*STI
6*      SX1 = -STHI
7*      SX2 = -STHF
8*      SY1=DEI(SX1)
9*      SY2=DEI(SX2)
10*     SA2= SY2-SY1
11*     SNF=SNI*EXP(-(SA1+SE/1.98717*SA2)*SB/SHR)
12*     RETURN
13*     END
```

## URNING RATE PREDICTION FOR MULTILAYERED AIRCRAFT SEAT CUSHION

MATERIAL PROPERTY	TEMP. RANGE (K)	THERMAL CONDUCTIV. (CAL/CM.S.C.)	HEAT CAPACITY (CAL/GH.C.)	PREEXP. FACTOR (GM/CM3)	ACTIVATION ENERGY (CAL/MOLE)	HEAT OF DECOMPOS. (CAL/G.M)	HEAT RATE (DEG C/SEC)
1 COVER FABR	848.-2	898.-2	.00010	.600	.213+05	29000.	.55
	813.-2	848.-2	.00010	.600	.506+07	-20000.	.55
2 VONAR	763.-2	813.-2	.00010	.600	.524+14	56500.	.55
	689.-2	763.-2	.00010	.600	.251+00	7000.	.55
3 LS 200	602.-2	669.-2	.00010	.600	.730+05	-6500.	.55
	523.-2	602.-2	.00010	.600	.157+03	13700.	.55
4 PREOX	881.-0	923.-0	.00020	.345	.146	.337+03	.13
	766.-0	881.-0	.00020	.345	.146	.166+06	.55
5 PU FOAM	709.-0	766.-0	.00020	.345	.146	.894+01	.55
	661.-0	709.-0	.00020	.345	.146	.237+11	.55
6 INIDE FOAM	623.-0	661.-0	.00020	.345	.146	.275+05	.55
	591.-0	623.-0	.00020	.345	.146	.275+05	.55
7 INIDE FOAM	473.-0	591.-0	.00020	.345	.146	.394+12	.55
	1006.-0	1073.-0	.00010	.400	.030	.444+20	.55
8 INIDE FOAM	973.-0	1006.-0	.00010	.400	.030	.617+02	.55
	915.-0	973.-0	.00010	.400	.030	.177+01	.55
9 INIDE FOAM	854.-0	896.-0	.00010	.400	.030	.260+06	.55
	711.-0	854.-0	.00010	.400	.030	.600+01	.55
10 INIDE FOAM	673.-0	711.-0	.00010	.400	.023	.306+04	.55
	1072.-0	1073.-0	.00010	.200	.023	.312+13	.55
11 INIDE FOAM	1073.-0	1073.-0	.00010	.200	.023	.180+06	.55
	1073.-0	1073.-0	.00010	.200	.023	.133+02	.55
12 INIDE FOAM	1073.-0	1073.-0	.00010	.200	.023	.611+05	.55
	1073.-0	1073.-0	.00010	.200	.023	.107+08	.55

## NORMALIZED FRAGMENT SIZE AT END TEMPS OF KINETIC REGIMES

## 1 COVER FAIR

523.2-	602.2	1.000000	--	.821322
602.2-	689.2	.821322	--	.602183
689.2-	763.2	.602183	--	.467861
763.2-	813.2	.467861	--	.193770
813.2-	848.2	.193770	--	.059566
848.2-	898.2	.059566	--	.007807

## 2 VONAR

473.0-	501.0	1.000000	--	.865850
591.0-	623.0	.865850	--	.753600
623.0-	661.0	.753600	--	.718568
661.0-	709.0	.718568	--	.644733
709.0-	766.0	.644733	--	.593173
766.0-	801.0	.593173	--	.535456
801.0-	823.0	.535456	--	.522000

## 3 LS 200

473.0-	548.0	1.000000	--	.967082
548.0-	573.0	.967082	--	.934163
573.0-	598.0	.934163	--	.471674
598.0-	623.0	.471674	--	.920223
623.0-	648.0	.920223	--	.781499
648.0-	673.0	.781499	--	.725286
673.0-	698.0	.725286	--	.684495
698.0-	723.0	.684495	--	.666312
723.0-	766.6	.666312	--	.615242
766.6-	876.6	.615242	--	.550394
876.6-	923.0	.550394	--	.542658

## 4 PRFOX

498.0-	573.0	1.000000	--	.953137
573.0-	623.0	.953137	--	.892503
623.0-	683.0	.892503	--	.829126
683.0-	748.0	.829126	--	.763548
748.0-	848.0	.763548	--	.542460
848.0-	873.0	.542460	--	.367852
873.0-	908.0	.367852	--	.146867

## 5 PU FOAM

448.0-	544.0	1.000000	--	.918883
544.0-	573.0	.918883	--	.834678
573.0-	598.0	.834678	--	.727952
598.0-	644.0	.727952	--	.490399
644.0-	688.0	.490399	--	.006318

## 6 IMIDE FOAM

673.0-	711.0	1.000000	--	.994404
711.0-	796.0	.994404	--	.917053
796.0-	854.0	.917053	--	.803908
854.0-	915.0	.803908	--	.679453
915.0-	973.0	.679453	--	.574808
973.0-	1006.0	.574808	--	.546754
1006.0-	1073.0	.546754	--	.513983

PURGING RATE PREDICTION FOR MULTI-LAYERED AIRCRAFT SEAT CUSHION

RUN NO. 1e 0.5 INCH LS-200 1±6 MIN

MATERIAL	SURFACE TEMP. (K)	Thermal Conductiv. (CAL/CM.SEC)	Heat Capacity (CAL/GM.C)	DENSITY (GM/CM <sup>3</sup> )	PREDICT. FACTOR	ACTIVATION ENERGY (EAL/POUL)	HEAT OF COMBUSTION (CAL/GM)	HEAT RATE (EFG C/SEC)
1 COVER FABR	691.0	.00010	.600	.400	.251*00	7800.	.5.	.333
2 COVER FABR	659.2	.00010	.600	.400	.733*05	-6500.	-16.	.333
3 LS 200	677.0	.00020	.300	.120	.828*13	-31043.	-17.	.333
4 LS 200	673.0	.00020	.300	.120	.656*02	2475.	-8.	.333
5 LS 200	644.0	.00020	.300	.120	.246*07	27697.	-20.	.333
6 LS 200	623.0	.00020	.300	.120	.161*16	-38100.	4.	.333
7 LS 200	598.0	.00020	.300	.120	.211*08	27800.	72.	.333
8 LS 200	573.0	.00020	.300	.120	.261*00	6772.	25.	.333
9 LS 200	544.0	.00020	.300	.120	.763*04	18279.	55.	.333
10 PU FOAM	459.0	.00010	.400	.030	.600*01	9900.	1.	.333
REAR SURF	291.							

PREDICTED BURNING RATE

NO OF ITERATION = 5	BURN RATE (CM/SEC)	SURFACE TEMP. (K)	HEAT FLUX IN (CAL/CM <sup>2</sup> SEC)	HEAT FLUX OUT (CAL/CM <sup>2</sup> SEC)	HEAT CONSUMED (CAL/CM <sup>2</sup> SEC)	FS-TOP SURFACE	FS-BOTTOM
1 COVER FABR	.00003	691.0	.0400	.0296	.0004	.24964*01	.25137*01
2 COVER FABR	.00084	687.2	.0396	.0372	.0024	.25137*01	.26443*01
3 LS 200	.00160	677.0	.0369	.0367	.0002	.35333*01	.36401*01
4 LS 200	.00161	673.0	.0367	.0352	.0015	.36401*01	.45766*01
5 LS 200	.00191	648.0	.0352	.0337	.0015	.45766*01	.55620*01
6 LS 200	.00194	623.0	.0337	.0318	.0018	.55624*01	.77926*01
7 LS 200	.00215	598.0	.0319	.0286	.0031	.77926*01	.15189*02
8 LS 200	.00234	573.0	.0286	.0264	.0023	.15189*02	.30370*02
9 LS 200	.00281	544.0	.0265	.0168	.0096	.30370*02	.10000*05
10 PU FOAM	.0036	459.0	.0174	.0004	.0168	.29784*01	.10000*05

WEIGHT LOSS PREDICTED BY THE PROFILE  
• DODGE GRAY/SEC.CM<sup>2</sup> OF BURNING SURFACE

```

1*      C      PROGRAM B
2*      C      THIS PROGRAM TAKES THE TEMPERATURE AND CONDUCTIVE HEAT FLUX AT THE
3*      C      FRONT (HOT) SURFACE (I.E., B.C.'S) AS THE INPUT DATA AND THEN CALCULATES
4*      C      THE BURNING RATE, TEMPERATURE PROFILE, AND WEIGHT LOSS OF MULTI-LAYERED
5*      C      POLYMERIC MATERIALS (INCLUDING FIRE BLOCKING LAYER) USED FOR AIRCRAFT
6*      C      SEAT CUSHIONS.
7*      C      THE DIMENSIONS USED IN THIS PROGRAM ARE GRAM, GRAM-MOLE, DEG-CELSIUS,
8*      C      CALORIE, CM, AND SECOND.
9*      C
10*      C      COMMON /TCM1/ANAMET(1,2),MW( 7),NTREG( 7),KT(7,11),CPT(7,11),
11*      C      1  PHOT(7,11),LT(7,11),ET(7,11),DT(7,11),HEATPT(7,11),TKLU(7,11),
12*      C      2  TKH(7,11),TS(2(1),N(201),FS(201),OS(201),FP(201),%2(201),
13*      C      3  NKL0(7,11),TKH1(7,11),FKL0(7,11),FKH1(7,11),ID(8),THICK(8),
14*      C      4  PHC(201),LA"PCA,R,I,J,I,TC,J1
15*      C      REAL KT, N, NKH1, NKL0, LAMRDA,MW
16*      C      DIMENSION HEADING(16),WLLOSS(7)
17*      C
18*      C      READ IN THE INGREDIENT MATERIAL PROPERTY AND PRINT OUT
19*      C
20*      C      READ (5,901) ITOTAL
21*      901 FORMAT (I2)
22*      DO 100 II=1,ITOTAL
23*      READ (5,902) I,ANAMET(I,1),ANAMET(I,2),MW(I),NTREG(I)
24*      902 FORMAT(I2,24F,10.2,12)
25*      NTK=NTREG(I)
26*      READ (5,903)(TKL0(I,J),TKH1(I,J),KT(I,J),CPT(I,J),PHOT(I,J),
27*      -          BT(I,J),ET(I,J),DT(I,J),HEATRT(I,J),J=1,NTR)
28*      903 FORMAT (2F5.1, 7E10.3)
29*      100 CONTINUE
30*      WRITE (6,944)
31*      944 FORMAT (1H1, 25X, *BURNING RATE PREDICTION FOR MULTI-LAYERED*
32*      - * AIRCRAFT SEAT CUSHION* / 26X, 63(*=*)+ /
33*      - 5X, *MATERIAL PROPERTY* / 21X, *TEMP. RANGE, THERMAL*, 5X,
34*      - *HEAT*, 5X, *DENSITY*, 5X, *ACTIVATION*, 5X, *HEAT*, 5X,
35*      - 5X, / 36X, *CONDUCTIV. CAPACITY*, 13X, *FACTOR ENERGY*, 5X,
36*      - 7X, *DECOMPOS. RATE*, 5X,
37*      - 26X, * (K)*, 5X, *(CAL/GM.S.C) (CAL/GM.C) (GR/GM3)*, 13X,
38*      - *(CAL/MOLE) (CAL/GM) (DEG C/SEC)* /)
39*      DO 102 I=1,ITOTAL
40*      NTRE=NTREG(I)
41*      WRITE (6,951) I,ANAMET(I,1),ANAMET(I,2)
42*      951 FORMAT (5X,I2,2X,24F)
43*      WRITE (6,953) (TKL0(I,J),TKH1(I,J),
44*      - KT(I,J),CPT(I,J),RHOT(I,J),RT(I,J),ET(I,J),DT(I,J),HEATRT(I,J),
45*      - J=1,NTK)
46*      953 FORMAT (17Y, F7.1, *-, F7.1, 1X,
47*      - F10.5,3Y, F6.3, 3Y, F6.3, 2Y, E11.3, 4X, F9.0, F10.1, F10.2)
48*      102 CONTINUE
49*      C
50*      C      FIX CONSTANTS USED IN THIS PROGRAM

```

```

51*      C
52*      F=1.0
53*      R=1.98717
54*      RM=82.06
55*      CHGAS=0.382
56*      EFSLDN=0.9
57*      SIGMA=1.355E-12
58*      TC=285.0
59*      C
60*      C CALCULATE N AND FS AT TWO END TEMPERATURES OF KINETIC REGIMES
61*      C THE CALCULATION SEQUENCE IS, BY LOGIC, FROM LOW TO HIGH TEMPERATURE
62*      C
63*      WRITE (6,955)
64*      955 FORMAT (1H1, 10X, *NORMALIZED FRAGMENT SIZE AT END TEMPS OF*, 1V,
65*      - *KINETIC REGIMES*, /, 11X, 56(*=*), /)
66*      DO 200 I=1,ITOTAL
67*          WRITE (6,951) I, ANAMET(I, 1), ANAMET(I,2)
68*          NTR=NTRREG(I)
69*          NKLO(I,NTR)=1.0
70*          FKLO(I,NTR)=10000.
71*          DO 205 JJ=1,NTR
72*              J=NTR+1-JJ
73*              CALL FRAGMT (HT(I,J),ET(I,J),TKLD(I,J),TKHI(I,J),
74*              - HEATPT(I,J),FKLO(I,J),NKHI(I,J))
75*              IF (J.NE.1) NKLO(I,J-1)=NKHI(I,J)
76*              IF (NKLO(I,J).EQ.1.0) NKLO(I,J)=0.9999999
77*              FKLO(I,J) = 1.0/(1.0-NKLO(I,J))
78*              IF (KKFI(I,J).EQ.1.0) KKFI(I,J)=0.9999999
79*              KKFI(I,J) = 1.0/(1.0-NKHI(I,J))
80*              WRITE (6,958) TKLO(I,J),TKHI(I,J),NKLO(I,J),NKHI(I,J)
81*      958 FORMAT (22V, F7.1, *-**, F7.1, 5V, F11.6, 2V, *--*, 1X, F11.6)
82*      205 CONTINUE
83*      200 CONTINUE
84*      C
85*      C INITIALIZE A FEW VARIABLES FOR COMPUTER PLOTTING
86*      C
87*      NPAGE=9
88*      NPLOT=1
89*      IPLOT=0
90*      CALL PGNPLT
91*      C
92*      C READ IN THE INPUT DATA, I.E., TOTAL NUMBER OF LAYERS AND THEIR ORDER
93*      C THICKNESS OF EACH LAYER (CM), AND FRONT SURFACE HEAT FLUX
94*      C AND FRONT SURFACE TEMPERATURE (K)
95*      C
96*      1112 CONTINUE
97*      C CLEAR THE MEMORIES OF PLOT
98*      DO 1336 I=1,201
99*          X(I)=0.0
100*      1338 TS(I)=0.0
101*      READ (5,907,END=1110) (HEADIN(I),I=1,16)
102*      907 FORMAT (16I5)
103*      READ (5,904) NLAYG,(ID(I),I=1,NLAYG)
104*      READ (5,905) (THICK(I),I=1,NLAYG)
105*      READ (5,906) QS1,TS1
106*      904 FORMAT (16I5)
107*      905 FORMAT (8F10.0)

```

```

18*      906 FORMAT(2F10.0)
19*      WRITE (6,955) (HEADIN(I),I=1,16)
20*      959 FORMAT (1H1,20X,16A5//)
21*      WRITE (6,960)
22*      960 FORMAT (10X, *MATERIAL / THICKNESS*,/)
23*      DO 250 I=1,NLAYG
24*          IDI=10(I)
25*      250 WRITE (6,961) ANAMET(IDI,1), AVAMET(IDI,2), THICK(I)
26*      961 FORMAT (20Y, 2A5, 5X, F5.1, * C*,/)
27*      WRITE (6,964) GS1,TS1
28*      964 FORMAT (10Y, *BOUNDARY CONDITION*,/, 20X, *HEAT FLUX (CONDUCTIVE)*
29*      - ,* =*, F10.4, * CAL/CM2 SEC*, /, 20X, *SURFACE TEMPERATURE =*,*
30*      - F6.1, * K*, /)
31*
32*      C
33*      C THE DIMENSIONS FOR TS, X, GS, NO RP HAVE TO BE LARGE ENOUGH TO COVER THE
34*      C WHOLE TEMPERATURE DROP FROM THE FRONT TO REAR SURFACE IS COVERED
35*      C WITH THE TEMPERATURE INTERVAL (TINTV) AS SHOWN BELOW
36*      C
37*      11,TV=4.0
38*      I=0
39*      TS(1)=TS1
40*      GS(1)=GS1
41*      V(1)=0.0
42*      STHICK=0.0
43*      TWLT=0.0
44*      DO 251 ING=1,7
45*      251 WLOSS(ING)=0.0
46*      WRITE (6,965)
47*      965 FORMAT (11X,*I*, 9Y, *RF*, 5Y), *TS(I) - TS(I+1)*, 8X, *D*,,
48*      - EX, *X(I) - V(I+1)*, 5Y,*GS(I) GS(I+1)*, 10X,*N(I) N(I+1)*/
49*      DO 500 III=1,NLAYG
50*          I1=I+1
51*          IDI=ID(III)
52*          WRITE (6,970) III, AVAMET(IDI,1), ANAMET(IDI,2)
53*          970 FORMAT (3Y, 13, 3X,2A5)
54*          NTR=NTREFG(IDI)
55*          STHICK=STHICK+THICK(III)
56*      502 CONTINUE
57*      I=I+1
58*      TS(I+1)=TS(I) - TINTV
59*      504 CONTINUE
60*      IF (TS(I+1).LT.T0) TS(I+1)=T0
61*      IF (TS(I).LE.TKLO(IDI,NTR)) V(I)=1.0
62*      IF (TS(I).LE.TKLO(IDI,NTR)) GO TO 310
63*      DO 307 J=1,NTR
64*      IF (TS(I).GT.TKLO(IDI,J).AND.TS(I+1).LE.TKH1(IDI,J)) GO TO 308
65*      307 CONTINUE
66*      308 J1=J
67*      IF (TS(I+1).LT.TKLO(IDI,J1)) TS(I+1)=TKLO(IDI,J1)
68*      CALL FRAGMT (FT(IDI,J),ET(IDI,J),TKLO(IDI,J),TS(I),
69*      1          HEATRT(IDI,J),NKLO(IDI,J),N(I))
70*      IF (N(I).EQ.1.0) FS(I)=10000.
71*      IF (N(I).NE.1.0) FS(I)=1.0/(1.0-N(I))
72*      CALL FRAGMT (FT(IDI,J),ET(IDI,J),TKLO(IDI,J),TS(I+1),
73*      1          HEATRT(IDI,J),NKLO(IDI,J),N(I+1))
74*      IF (N(I+1).EQ.1.0) FS(I+1)=10000.
75*      IF (N(I+1).NE.1.0) FS(I+1)=1.0/(1.0-N(I+1))

```

```

65*      C
66*      C CALCULATE THE BURNING RATE R USING THE HEAT FLUX AT THE FRONT
67*      C AND THE ASSUMED HEAT FLUX (VIA REAR SURFACE TEMPERATURE)
68*      C
69*      CALL BURNRT
70*      RHO(I)=RHOT(IDI,J1)
71*      DTDX=QS(I)/KT(IDI,J1)
72*      DX=(TS(I)-TS(I+1))/DTDX
73*      GO TO 405
74* 310 CONTINUE
75*      IF (TS(I+1).LE.TKLO(IDI,NTR)) N(I+1)=1.0
76*      DENOM=RHOT(IDI,NTR)*CPT(IDI,NTR)*(TS(I)-T0)
77*      RF(I)=QS(I)/DENOM
78*      C=KT(IDI,NTF)*(TS(I))-TS(I+1))/(DENOM*RF(I))
79*      RHO(I)=RHOT(IDI,NTR)
80*      QS(I+1)=QS(I)-RHOT(IDI,NTR)*RF(I)*CPT(IDI,NTR)*(TS(I)-TS(I+1))
81* 405 CONTINUE
82*      X(I+1)=X(I)+DX
83*      IF (X(I+1)/STHICK-1.0).LT.0.01 GO TO 499
84*      IF (X(I+1)).LE.1.0E-04 GO TO 499
85*      IF (X(I+1)/T0-1.0).LT.0.0035 GO TO 499
86*      IF (X(I+1).LT.STHICK-.4*DX,QS(I+1).LT.0.0) WRITE (6,972) I,RF(I),
87*      - TS(I),TS(I+1),DX,Y(I+1),QS(I),QS(I+1),N(I+1)
88* 572 FORMAT(1CX, I3, F10.5, 3X, F6.1, ' --', F6.1, 5X, F6.2, F8.4, ' --',
89*      - F8.4, 2', F10.4, F10.4, 2', F10.5,F10.5)
90*      IF (Y(I+1).LT.STHICK.AND.QS(I+1).GT.0.0) GO TO 502
91*      IF (Y(I+1).GE.STHICK) PFACT=(STHICK-(I+1))/DX
92*      IF (QS(I+1).LE.0.0) PFACT=QS(I)/(QS(I)-QS(I+1))
93*      TS(I+1)=TS(I)-(TS(I))-TS(I+1))*PFACT
94*      GO TO 504
95* 499 CONTINUE
96*      WRITE (6,972) I,RF(I),TS(I),TS(I+1),
97*      - D/,X(I),X(I+1),QS(I),QS(I+1),N(I),N(I+1)
98*      C ***NOTE*** N(I+1) OF ABOVE WILL BE REPLACED BY THE FIRST OF THE NEXT LAY
99*      I2=I
100*      DC 600 J=11,12
101*      WLT=RF(J)*RHO(J)*(I(J+1)-I(J))
102*      IF (WLT.LT.0.0) WRITE (6,976) J,WLT,WLOSS(IJ)
103* 576 FORMAT(1CX, 'WLT', I3, ' =', F10.5, ' WLOSS =', F10.5)
104*      600 WLOSS(IJ)=WLOSS(IJ)+WLT
105*      TWLT=TWLT+WLOSS(IJ)
106* 500 CONTINUE
107*      WRITE (6,982) (IJ,WLOSS(IJ),IJ=1,LAY)
108* 982 FORMAT(1CX, 'MASS LOSS FOR ', I3, ' LAYER =', E11.4,/
109*      - (24Y, I3, ' LAYER =', E11.4/))
110*      WRITE (6,977) TWLT
111* 977 FORMAT(2X,5E0.0, ' TOTAL MASS LOSS =', F11.4, ' GM/SEC.CM2 OF '
112*      - 'BURNING SURFACE')
113*      IPLOT=IPLOT+1
114*      IF (IPLOT.GT.5) IPLOT=1
115*      IF (IPLOT.EQ.1.AND.NPLOT.NE.NPAGE) CALL ADVPLT
116*      CALL PLFORM ('L1LIN',2.5,2.75)
117*      CALL PLAREL ('TEMPERATURE PROFILE', 0,
118*      - 'DEPTH FROM SURFACE (CM)', 0, 'TEMPERATURE (K)', 0)
119*      CALL PLSCAL ((0.0,10.0)+2.100000*(300.0,400.0)+2.100000)
120*      CALL PLCS17(0.06,0.0)
121*      GO TO (1341+1342+1343+1344+1345), IPLOT

```

```

22*      1341 CALL ORIGIN (1.0,4.3)
23*          GO TO 134A
24*      1342 CALL ORIGIN (3.0,0.0)
25*          GO TO 134A
26*      1343 CALL ORIGIN (3.0,0.0)
27*          GO TO 134A
28*      1344 CALL ORIGIN (-6.0,-3.1)
29*          GO TO 134A
30*      1345 CALL ORIGIN (3.0,0.0)
31*          NPLCT=NPLCT+1
32*      134P CALL PLGPAP
33*          CALL PLCURV (X,TS,201,0,"")
34*          GO TO 1112
35* 1310 CONTINUE
36*          CALL ENDPLT
37*          STOP
38*          END

```

```

1*      SUBROUTINE FRACHT(SH,SE,STI,STF,SHR,SHI,SAF)
2*      DOUBLE PRECISIONN SX1,SX2,SY1,SY2,DLI
3*      STH=SE/1.96717*STI
4*      STHF=SE/(1.96717*STF)
5*      SA1=EXP(-STHF)*STF - EXP(-STH)*STI
6*      SY1 = -STH
7*      SY2 = -STHF
8*      SY1=DFI(SY1)
9*      SY2=DFI(SY2)
10*     SA2= SY2-SY1
11*     SAF=SHI*SE/((SA1+SE/1.96717*SA2)*SH/SHR)
12*     RETURN
13*     END

```

```

1*      SUBROUTINE BURART
2*      COMMON /TCM1/ANAME(7,2),MW( 7),NTREG( 7),KT(7,11),CPT(7,11),
3*      1 RHO(7,11),ET(7,11),DT(7,11),H5ATPT(7,11),TKL(7,11),
4*      2 TKH(7,11),TS(201),L(201),FS(201),JS(201),RR(201),X(201),
5*      3 NKL(7,11),NKH(7,11),NKL(7,11),EKH(7,11),ID(9),THICK(8),
6*      4 RHO(201),LAPRDA,R,I,IDI,T0,JI
7*      REAL KT, N, NKH, NKL, LAPRDA,MW
8*      IF (I.EQ.1) RF(1)=0.001
9*      IF (I.NE.1) RR(I)=RR(I)-1
10*     IF (N(I).EQ.1.0) GO TO 378
11* 377 CONTINUE
12*     PHT=RR(I)
13*     PS=-QS(I)/((TS(I)-T0)+RF(I)+RHO(IDI,J1)+CPT(IDI,J1))
14*     H=DT(IDI,J1)/(CPT(IDI,J1)*(TS(I)-T0))
15*     G=PS-1.0+H*N(I)
16*     THETA=ET((IDI,J1)/(P*TS(I)))
17*     ETAC=(ET((IDI,J1)/R)*(TS(I)-TS(I+1))/(TS(I)+TS(I)))

```

```

18*      ENC=EXP(-ETAC)
19*      CHI=(TS(I)-TC)/TS(I)
20*      ALPHAE=ET(I,J1)/(RHOT(I,J1)*CFT(I,J1))
21*      LAMDA=(H*(N(I)-N(I+1))+(1.0+0)*ALOG(N(I+1)/N(I)))/N(I-1)
22*      RSG=ALPHA*FT(I,J1)*EXP(-THETA)/(LAMDA*THETA*CHI)
23*      IF (RSG.LT.0.0) WRITE (6,910) I,TS(I),TS(I+1),RSG
24*      910 FORMAT (3X,'I=',I3, ' TS(I)=',F6.1, ' TS(I+1)=',F6.1, ' RSG=',0.
25*                      1 E10.4)
26*      IF (RSG.LT.0.0) RSG=-RSG
27*      FF(I)=SQRT(RSG)
28*      IF (4.05*(RHT/RG(I)-1.0).GT.0.005) GO TO 377
29*      376 CONTINUE
30*      TS(I+1)=GS(I)-RG(I)*RHOT(I,J1)*(CFT(I,J1)*(TS(I)-TS(I+1))
31*                      1 *ET(I,J1)*(N(I+1)-N(I)))
32*      RE TURN
33*      END

```

## PUPPYING RATE PRECISION FOR MULTI-LAYERED AIRCRAFT GEAR CUSHION

MATERIAL PROPERTY		TEMP. RANGE.		THERMAL CONDUCTIV.		HEAT CAPACITV.		INTENSY.		PREFTR. FACTOR		ACTIVATION ENERGY		HEAT OF DECOMPOS.		HEAT RATE (DEG C/SFC)	
		(K)		(WAL/C4.SFC)	(WAL/C4.SFC)	(CAL/60.C)	(J60/60.C)	(W/m.C)	(W/m.C)	(W/m.C)	(W/m.C)	(eV)	(eV)	(CAL/100)	(CAL/100)	(CAL/100)	(CAL/100)
1	COVER FAIR	840.0-	980.0	• 00010	• 100	• 400	• 213•05	24800•	• 213•05	24800•	• 213•05	24800•	24800•	24800•	-2.9	• 53	• 53
		A13.2-	880.0	• 00100	• 300	• 400	• 506•07	-20300•	• 506•07	-20300•	• 506•07	-20300•	-20300•	-20300•	-2.9	• 53	• 53
		763.2-	913.0	• 00010	• 300	• 400	• 229•14	• 6500•	• 6500•	• 6500•	• 6500•	• 6500•	• 6500•	• 6500•	-2.9	• 53	• 53
		609.2-	763.0	• 00010	• 300	• 400	• 251•00	7400•	7400•	7400•	7400•	7400•	7400•	7400•	-2.4	• 53	• 53
		607.0-	687.0	• 00010	• 300	• 400	• 730•05	• 6500•	• 6500•	• 6500•	• 6500•	• 6500•	• 6500•	• 6500•	-16.3	• 53	• 53
2	VOLAP	523.0-	602.0	• 00010	• 300	• 400	• 157•03	15700•	15700•	15700•	15700•	15700•	15700•	15700•	-16.3	• 53	• 53
		A11.0-	923.0	• 00020	• 145	• 146	• 337•03	1043•	1043•	1043•	1043•	1043•	1043•	1043•	1.0	• 53	• 53
		766.0-	861.0	• 00020	• 345	• 146	• 165•06	-12245•	• 9.6	-12245•	• 9.6	-12245•	-12245•	-12245•	-23.9	• 53	• 53
		705.0-	766.0	• 00020	• 345	• 146	• 699•01	7600•	7600•	7600•	7600•	7600•	7600•	7600•	-23.9	• 53	• 53
		661.0-	709.0	• 00020	• 345	• 146	• 237•11	-26532•	-26532•	-26532•	-26532•	-26532•	-26532•	-26532•	-26.7	• 53	• 53
		620.0-	661.0	• 00020	• 345	• 146	• 275•05	22079•	22079•	22079•	22079•	22079•	22079•	22079•	-26.7	• 53	• 53
		591.0-	623.0	• 00020	• 345	• 146	• 399•32	-25933•	-25933•	-25933•	-25933•	-25933•	-25933•	-25933•	-50.9	• 53	• 53
		473.0-	561.0	• 00020	• 345	• 146	• 233•02	19435•	19435•	19435•	19435•	19435•	19435•	19435•	107.5	• 53	• 53
3	LS 200	A76.6-	923.0	• 00020	• 300	• 120	• 117•04	-34557•	-34557•	-34557•	-34557•	-34557•	-34557•	-34557•	9.6	• 53	• 53
		756.6-	876.6	• 00020	• 300	• 120	• 717•10	-24770•	-24770•	-24770•	-24770•	-24770•	-24770•	-24770•	-23.9	• 53	• 53
		723.0-	766.6	• 00020	• 300	• 120	• 517•02	16816•	16816•	16816•	16816•	16816•	16816•	16816•	-8.4	• 53	• 53
		689.0-	723.0	• 00020	• 300	• 120	• 104•03	-3020•	-3020•	-3020•	-3020•	-3020•	-3020•	-3020•	-19.1	• 53	• 53
		613.0-	698.0	• 00020	• 300	• 120	• 429•13	-31043•	-31043•	-31043•	-31043•	-31043•	-31043•	-31043•	-16.7	• 53	• 53
		646.0-	673.0	• 00020	• 300	• 120	• 656•02	2475•	2475•	2475•	2475•	2475•	2475•	2475•	-8.4	• 13	• 13
		623.0-	646.0	• 00020	• 300	• 120	• 246•07	27497•	27497•	27497•	27497•	27497•	27497•	27497•	-20.3	• 53	• 53
		599.0-	623.0	• 00020	• 300	• 120	• 161•16	-38160•	-38160•	-38160•	-38160•	-38160•	-38160•	-38160•	7.9	• 53	• 53
		573.0-	599.0	• 00020	• 300	• 120	• 211•02	27400•	27400•	27400•	27400•	27400•	27400•	27400•	71.7	• 53	• 53
		549.0-	573.0	• 00020	• 300	• 120	• 201•00	6772•	6772•	6772•	6772•	6772•	6772•	6772•	24.9	• 53	• 53
		473.0-	548.0	• 00020	• 300	• 120	• 783•04	18279•	18279•	18279•	18279•	18279•	18279•	18279•	54.9	• 53	• 53
4	PTOK	A13.0-	908.0	• 00030	• 300	• 620	• 542•05	-13061•	-13061•	-13061•	-13061•	-13061•	-13061•	-13061•	.5	• 53	• 53
		A49.0-	A73.0	• 00030	• 300	• 620	• 655•21	91578•	91578•	91578•	91578•	91578•	91578•	91578•	.5	• 53	• 53
		74F.0-	64R.0	• 00030	• 300	• 620	• 155•03	14860•	14860•	14860•	14860•	14860•	14860•	14860•	-.5	• 53	• 53
		683.0-	748.0	• 00030	• 300	• 620	• 585•02	4500•	4500•	4500•	4500•	4500•	4500•	4500•	-13.9	• 53	• 53
		625.0-	663.0	• 00030	• 300	• 620	• 143•04	-4365•	-4365•	-4365•	-4365•	-4365•	-4365•	-4365•	-13.9	• 53	• 53
		573.0-	623.0	• 00030	• 300	• 620	• 587•02	3085•	3085•	3085•	3085•	3085•	3085•	3085•	-13.9	• 53	• 53
		49H.0-	573.0	• 00030	• 300	• 620	• 574•01	10963•	10963•	10963•	10963•	10963•	10963•	10963•	.5	• 53	• 53
5	PU FOAM	644.0-	688.0	• 00010	• 600	• 030	• 644•20	65757•	65757•	65757•	65757•	65757•	65757•	65757•	-3.1	• 53	• 53
		595.0-	644.0	• 00010	• 600	• 030	• 617•02	12335•	12335•	12335•	12335•	12335•	12335•	12335•	-11.9	• 53	• 53
		573.0-	595.0	• 00010	• 600	• 030	• 179•01	2646•	2646•	2646•	2646•	2646•	2646•	2646•	2.5	• 53	• 53
		548.0-	573.0	• 00010	• 600	• 030	• 260•06	21435•	21435•	21435•	21435•	21435•	21435•	21435•	2.5	• 53	• 53
6	IMIDE FCAP	1006.0-	1073.0	• 00010	• 200	• 023	• 106•04	-4763•	-4763•	-4763•	-4763•	-4763•	-4763•	-4763•	.5	• 53	• 53
		973.0-	1006.0	• 00010	• 200	• 023	• 312•13	-46170•	-46170•	-46170•	-46170•	-46170•	-46170•	-46170•	-.5	• 53	• 53
		915.0-	973.0	• 00010	• 200	• 023	• 1P0•06	• 1P0•06	• 1P0•06	• 1P0•06	• 1P0•06	• 1P0•06	• 1P0•06	-.5	• 53	• 53	
		874.0-	915.0	• 00010	• 200	• 023	• 135•02	16862•	16862•	16862•	16862•	16862•	16862•	16862•	-7.1	• 53	• 53
		795.0-	874.0	• 00010	• 200	• 023	• 611•03	-7869•	-7869•	-7869•	-7869•	-7869•	-7869•	-7869•	-7.3	• 53	• 53
		711.0-	796.0	• 00010	• 200	• 023	• 107•04	36705•	36705•	36705•	36705•	36705•	36705•	36705•	-7.0	• 53	• 53
		673.0-	711.0	• 00010	• 200	• 023	• 264•02	36705•	36705•	36705•	36705•	36705•	36705•	36705•	-7.0	• 53	• 53

NORMALIZED FRAGMENT SIZE AT END TEMPS OF KINETIC REGIMES

=====

1 COVER FABR

523.2-	602.2	1.000000	--	.821322
602.2-	669.2	.821322	--	.602183
669.2-	763.2	.602183	--	.467861
763.2-	813.2	.467861	--	.193770
813.2-	848.2	.193770	--	.059566
848.2-	896.2	.059566	--	.007807

2 VONAR

473.0-	591.0	1.000000	--	.865850
591.0-	623.0	.865850	--	.793600
623.0-	661.0	.793600	--	.718560
661.0-	709.0	.718560	--	.644733
709.0-	766.0	.644733	--	.598173
766.0-	881.0	.598173	--	.535456
881.0-	923.0	.535456	--	.522900

3 LS 200

473.0-	548.0	1.000000	--	.967082
548.0-	573.0	.967082	--	.934163
573.0-	598.0	.934163	--	.871674
598.0-	623.0	.871674	--	.820223
623.0-	648.0	.820223	--	.781499
648.0-	673.0	.781499	--	.725286
673.0-	698.0	.725286	--	.688495
698.0-	723.0	.688495	--	.666312
723.0-	766.0	.666312	--	.615242
766.0-	876.0	.615242	--	.550394
876.0-	923.0	.550394	--	.542658

4 PREOX

498.0-	573.0	1.000000	--	.953137
573.0-	623.0	.953137	--	.892503
623.0-	683.0	.892503	--	.826126
683.0-	748.0	.826126	--	.763548
748.0-	848.0	.763548	--	.592460
848.0-	873.0	.592460	--	.367852
873.0-	908.0	.367852	--	.146867

5 PU FOAM

448.0-	544.0	1.000000	--	.913843
544.0-	573.0	.913843	--	.834678
573.0-	598.0	.834678	--	.727952
598.0-	644.0	.727952	--	.490399
644.0-	698.0	.490399	--	.006318

6 IMPIDE FOAM

673.0-	711.0	1.000000	--	.994404
711.0-	796.0	.994404	--	.917053
796.0-	854.0	.917053	--	.803906
854.0-	915.0	.803906	--	.679453
915.0-	973.0	.679453	--	.574808
973.0-	1006.0	.574808	--	.546754
1006.0-	1073.0	.546754	--	.513953

RUN NO. 10

0.5-14CH LS-2CC (T=1.411)

## MATERIAL / THICKNESS

COPRA FABR

.1 CM

.5 200

1.0 CM

PU FOAM

10.0 CM

BOUNDARY CONDITION  
HEAT FLUX (CONDUCTIVE) = 0000 CAL/CM<sup>2</sup> SEC  
SURFACE TEMPERATURE = 691.0 K

			1	RB	15(1) - TS(1+1)	DX	Y(1) - Y(1+1)	05(1)	05(1+1)	N(1)	N(1+1)	
1												
1	COVER FABR				691.0 -- 699.2	.00	.0000-	.3045	.0430	.59442	.60218	
2		.00084	691.0 -- 695.2	.01	.0145-	.0145	.0390	.0395	.0395	.60837	.60837	
3		.00084	685.2 -- 691.2	.01	.0145-	.3247	.0395	.0395	.0395	.61420	.61420	
4		.00085	681.2 -- 677.2	.01	.0247-	.0349	.0391	.0391	.0391	.62167	.62167	
5		.00086	677.2 -- 673.2	.01	.0349-	.0452	.0391	.0391	.0391	.62849	.62849	
6		.00087	673.2 -- 669.2	.01	.0452-	.0557	.0383	.0383	.0383	.63564	.63564	
7		.00087	669.2 -- 665.2	.01	.0557-	.0662	.0380	.0380	.0380	.64323	.64323	
8		.00088	665.2 -- FB1.2	.01	.0662-	.0769	.0376	.0376	.0376	.65103	.65103	
9		.00089	661.2 -- 657.2	.01	.0769-	.0876	.0372	.0372	.0372	.65109	.65109	
10		.00090	657.2 -- 653.2	.01	.0876-	.0985	.0368	.0368	.0368	.65929	.65929	
11		.00091	653.2 -- 652.6	.00	.0985-	.1000	.0364	.0364	.0364	.66904	.66904	
2												
12		.00103	652.6 -- 649.6	.02	.1000-	.1220	.0354	.0354	.0354	.7106	.7106	
13		.00104	649.6 -- 649.0	.00	.1220-	.1255	.0361	.0361	.0361	.74006	.74006	
14		.00104	648.0 -- 644.0	.02	.1255-	.1477	.0361	.0361	.0361	.74997	.74997	
15		.00105	644.0 -- 640.0	.02	.1477-	.1700	.0357	.0357	.0357	.74744	.74744	
16		.00106	640.0 -- 636.0	.02	.1700-	.1924	.0357	.0357	.0357	.80401	.80401	
17		.00107	636.0 -- 632.0	.02	.1924-	.2150	.0354	.0354	.0354	.80974	.80974	
18		.00108	632.0 -- 628.0	.02	.2150-	.2378	.0352	.0352	.0352	.81482	.81482	
19		.00109	628.0 -- 624.0	.02	.2378-	.2507	.0349	.0349	.0349	.8150	.8150	
20		.00109	624.0 -- 623.6	.01	.2507-	.2664	.0347	.0347	.0347	.81322	.81322	
21		.00110	623.0 -- 619.0	.02	.2664-	.2864	.0346	.0346	.0346	.82022	.82022	
22		.00110	619.0 -- 615.0	.02	.2864-	.3129	.0345	.0345	.0345	.82460	.82460	
23		.00110	615.0 -- 611.0	.02	.3129-	.3364	.0340	.0340	.0340	.82994	.82994	
24		.00117	611.0 -- 607.0	.02	.3364-	.3601	.0337	.0337	.0337	.83663	.83663	
25		.00119	607.0 -- 603.4	.02	.3601-	.3840	.0334	.0334	.0334	.84497	.84497	
26		.00201	603.0 -- 599.0	.02	.3840-	.4082	.0331	.0331	.0331	.85513	.85513	
27		.00203	599.0 -- 595.0	.01	.4082-	.4143	.0327	.0327	.0327	.86797	.86797	
28		.00204	598.0 -- 594.0	.02	.4143-	.4347	.0327	.0327	.0327	.87167	.87167	
29		.00207	594.0 -- 590.0	.02	.4347-	.4635	.0322	.0322	.0322	.87167	.87167	
30		.00210	590.0 -- 586.0	.03	.4635-	.4908	.0317	.0317	.0317	.87595	.87595	
31		.00214	586.0 -- 582.0	.03	.4908-	.5145	.0312	.0312	.0312	.88011	.88011	
32		.00217	582.0 -- 578.0	.03	.5145-	.5405	.0317	.0317	.0317	.88484	.88484	
33		.00220	578.0 -- 574.0	.03	.5405-	.5610	.0317	.0317	.0317	.88597	.88597	
34		.00223	574.0 -- 573.0	.01	.5610-	.5717	.0294	.0294	.0294	.893256	.893256	
35		.00224	573.0 -- 569.0	.03	.5717-	.6004	.0291	.0291	.0291	.89416	.89416	
36		.00227	569.0 -- 565.0	.03	.6004-	.6274	.0291	.0291	.0291	.89496	.89496	
37		.00229	565.0 -- 561.0	.03	.6274-	.6555	.0291	.0291	.0291	.89575	.89575	
38		.00232	561.0 -- 557.0	.03	.6555-	.6835	.0291	.0291	.0291	.89612	.89612	
39		.00235	557.0 -- 553.0	.03	.6835-	.7114	.0291	.0291	.0291	.89611	.89611	

PU FOAM		553.0	549.0	0.03	7118-	7405	0.0219	965911
40	0.00238	549.0	546.0	0.01	7405-	7478	0.0275	967048
41	0.00242	546.0	544.0	0.01	7478-	7474	0.0275	967131
42	0.00243	546.0	544.0	0.01	7474-	7474	0.0275	967038
43	0.00246	544.0	540.0	0.03	7770-	.P067	0.0270	975348
44	0.00251	540.0	536.0	0.03	.B167-	.A369	0.0265	971446
45	0.00254	536.0	532.0	0.03	.A368-	.B165	0.0261	975348
46	0.00259	532.0	528.0	0.03	.B165-	.A366	0.0257	971446
47	0.00263	528.0	524.0	0.03	.A366-	.B167	0.0253	971446
48	0.00268	524.0	520.0	0.03	.B167-	.A369	0.0249	98455
49	0.00272	520.0	516.0	0.03	.A369-	.B165	0.0244	98455
50	0.00277	516.0	512.0	0.03	.B165-	.A368	0.0240	98455
51	0.00282	512.0	508.0	0.03	.A368-	.B167	0.0236	992356
52	0.00287	508.0	504.0	0.03	.B167-	.A369	0.0232	992356
53	0.00293	504.0	500.0	0.04	.A368-	.B165	0.0223	99373
54	0.00299	500.0	496.0	0.04	.B165-	.A369	0.0227	99595
55	0.00305	496.0	492.0	0.04	.A369-	.B167	0.0223	99595
56	0.00311	492.0	488.0	0.04	.B167-	.A368	0.0218	99683
57	0.00318	488.0	484.0	0.04	.A368-	.B165	0.0214	99683
58	0.00325	484.0	480.0	0.04	.B165-	.A369	0.0209	99827
59	0.00333	480.0	476.0	0.04	.A369-	.B167	0.0205	99883
60	0.00172	476.0	472.0	0.02	1.3559-	1.3799	0.0195	99932
61	0.00172	472.0	468.0	0.02	1.3799-	1.4005	0.0194	99932
62	0.00172	468.0	464.0	0.02	1.4005-	1.4212	0.0193	99140
63	0.00173	464.0	460.0	0.02	1.4212-	1.4420	0.0192	99140
64	0.00173	460.0	456.0	0.02	1.4420-	1.4628	0.0192	99629
65	0.00174	456.0	452.0	0.02	1.4628-	1.4836	0.0191	99765
66	0.00175	452.0	448.0	0.02	1.4836-	1.5044	0.0190	99888
67	0.00967	448.0	444.0	0.02	1.5044-	1.5260	0.0189	1.00000
68	0.00967	444.0	440.0	0.02	1.5260-	1.5477	0.0189	99480
69	0.00967	440.0	436.0	0.02	1.5477-	1.5700	0.0180	99629
70	0.00967	436.0	432.0	0.02	1.5700-	1.5928	0.0175	99765
71	0.00967	432.0	428.0	0.02	1.5928-	1.6162	0.0171	99888
72	0.00967	428.0	424.0	0.02	1.6162-	1.6404	0.0171	99888
73	0.00967	424.0	420.0	0.02	1.6404-	1.6652	0.0166	1.00000
74	0.00967	420.0	416.0	0.03	1.6652-	1.6907	0.0161	99140
75	0.00967	416.0	412.0	0.03	1.6907-	1.7170	0.0157	99140
76	0.00967	412.0	408.0	0.03	1.7170-	1.7442	0.0152	99629
77	0.00967	408.0	404.0	0.03	1.7442-	1.7722	0.0147	99765
78	0.00967	404.0	400.0	0.03	1.7722-	1.8012	0.0143	99888
79	0.00967	400.0	396.0	0.03	1.8012-	1.8312	0.0133	1.00000
80	0.00967	396.0	392.0	0.03	1.8312-	1.8622	0.0133	99140
81	0.00967	392.0	388.0	0.03	1.8622-	1.8945	0.0124	99140
82	0.00967	388.0	364.0	0.03	1.8945-	1.9279	0.0119	99888
83	0.00967	364.0	360.0	0.03	1.9279-	1.9626	0.0115	1.00000
84	0.00967	360.0	356.0	0.03	1.9626-	1.9911	0.0110	99140
85	0.00967	356.0	352.0	0.03	1.9911-	2.0370	0.0106	99140
86	0.00967	352.0	348.0	0.04	2.0370-	2.0766	0.0101	99765
87	0.00967	348.0	344.0	0.04	2.0766-	2.1141	0.0096	99888
88	0.00967	344.0	360.0	0.04	2.1141-	2.1618	0.0092	1.00000
89	0.00967	360.0	356.0	0.05	2.1618-	2.2078	0.0087	1.00000
90	0.00967	356.0	352.0	0.05	2.2078-	2.2563	0.0082	1.00000
91	0.00967	352.0	348.0	0.05	2.2563-	2.3078	0.0078	1.00000
92	0.00967	348.0	344.0	0.05	2.3078-	2.3625	0.0073	1.00000
93	0.00967	344.0	340.0	0.06	2.3625-	2.4210	0.0068	1.00000
94	0.00967	340.0	336.0	0.06	2.4210-	2.4837	0.0064	1.00000
95	0.00967	336.0	332.0	0.07	2.4837-	2.5513	0.0059	1.00000

90	.00967	332.0	--	328.0	.07	2.5513-	2.6246-	1.30030
97	.00967	328.0	--	329.0	.09	2.6246-	2.7046	1.00000
96	.00967	324.0	--	320.0	.09	2.7048-	2.7932	1.00000
99	.00967	326.0	--	316.0	.10	2.7932-	2.8917	1.00000
100	.00967	316.0	--	312.0	.11	2.8917-	3.0030	1.00000
101	.00967	312.0	--	304.0	.13	3.0030-	3.1307	1.00000
102	.00967	304.0	--	304.0	.15	3.1307-	3.2806	1.00000
103	.00967	304.0	--	303.0	.1H	3.2806-	3.4621	1.00000
104	.00967	300.0	--	296.0	.23	3.4621-	3.6926	1.00000
105	.00967	296.0	--	292.0	.31	3.6926-	4.0054	1.00000
106	.00967	292.0	--	288.0	.49	4.0054-	4.4980	1.00000
107	.00967	288.0	--	285.0	.86	4.4980-	5.3600	1.00000
PASS LOSS FOR		1 LAYER	--	2420-04				
PASS LOSS FOR		2 LAYER	--	.5P51-04				
3 LAYER		--	.5464-06					

\*\*\*\*\* TOTAL MASS LOSS = .4323-04 GR/SEC.CM<sup>2</sup> OF TURNING SURFACE

## APPENDIX C Some Comments on Burning Tests Procedure

The burning tests of multi-layered seat cushion assemblies were conducted in an NBS Smoke Density Chamber. The original radiative heat source is replaced by a Mellen furnace Model 10, to provide a high heat flux, and further supplemented with a cantilever beam coupled to a transducer, to measure weight changes of suspended samples.

In addition to these, there were several modifications on the procedures of sample preparation and its burning. These were instated following the detection of unsatisfactory conditions during the initial trial runs. These are summarized below.

Symptom	Diagnosis	Remedy
A) Uneven heating (Severe 2-D effect)	<ol style="list-style-type: none"> <li>1. Misalignment of sample with furnace</li> <li>2. Use of stainless steel frame as sample holder</li> <li>3. Loss of heat on four sides</li> <li>4. Buoyancy of hot gas formed in foam (open cell)</li> </ol>	<ol style="list-style-type: none"> <li>1. Align correctly</li> <li>2. Eliminate the frame, i.e., large heat sink item</li> <li>3. Use of light, thick insulator on four sides, also as sample holder</li> <li>4. No remedy unless sample positioned horizontally</li> </ol>
B) Irregular surface shape  (or disengaging the cover sheet in the worst case)	High pressure exerted by the gas product during intumescence	Cover fabric cut larger than the surface area, folded over the sides and stapled
C) Temperature rise at the sample surface before removing	Single sheet of radiation shield -- not enough	Use double layers (with air-gap in between) of shields

Symptom	Diagnosis	Remedy
D) Furnace temperature change on and after removing the shield	Change in Characteristics of reradiation from the shield and the sample surface	Adjust the power to the furnace manually to maintain the same furnace temperature
E) Non-linear signal response of transducer	Electronic feature of the equipment used	Make and use a calibration chart
F) Irreproducible weight measurement of sample	Irregularly distributed loading on the cantilever beam	Hold the entire sample by a single wire and load it at the fixed point of the beam
G) Step changes in weight measurement during the burning	Interference by the thermocouples inserted into the sample	Duplicate the run without thermocouples purely for weight change measurement

**APPENDIX D      Photographs of Equipments and Burned Samples**

<u>Photo No.</u>	<u>Remarks</u>
1. <u>Thermogravimetric Analyzer (TG)</u>	The thermogravimetric analyzer is shown in the right side of the picture, connected to a pair of flowmeters and a mixer, plus a vacuum gauge (vacuum pump not shown). The assemblage of a chart recorder, a microprocessor, a heater controller, and a weight suppressor are shown in the left side on the table.
2. <u>Differential Scanning Calorimeter (DSC)</u>	The DSC is shown here with a SAZ (Scanning Auto Zero) on its top. The sample heater unit is located on the deck on the left side. The DSC is not used extensively in this work.
3. <u>NBS Smoke Density Chamber</u>	The NBS Smoke Density Chamber is shown at left rear. A newly installed Mellen furnace is seen through its window. The multichannel recorder (Leeds & Northrop, Speedomax Model 251) is shown at the top of the front cabinet.
4. <u>Sample Preparation</u>	This picture shows each seat cushion sample is prepared for the burning tests in the NBS Smoke Density Chamber. The front cover fabric is cut larger than the face area and folded over the sides and stapled. The entire sample is surrounded by four 1"-thick insulator panels (Kaewood-3000° Board, by Pyroengineering Co.).
5. <u>Sample Arrangement Inside of the NBS Smoke Density Chamber</u>	Eight pairs of thermocouple (Pt vs. Pt-10% Rh) wires are shown here penetrating into the seat cushion assemblage, which is suspended by a cantilever beam. The beam is a part of a transducer-type weight measuring device. The Mellen furnace is shown at the lower left corner.
6. <u>Visual Comparison of the Burned Seat Cushions</u>	Each seat cushion system was suddenly exposed to the Mellen furnace preheated and maintained at 850°C and then remained so for 10 minutes. This picture shows that the effectiveness of the fire blocking layer can be rated, based on this test only, in decreasing order: LS-200 (1/2" thick) > Vonar (3/16" thick) > Preox (11 oz/yd <sup>2</sup> ) > No blocking layer.
7. <u>Comparison of the Degree of In-Depth Burning</u>	Same as above, but this picture shows the degree of in-depth burning, with the samples oriented in the same way as in the NBS Smoke Density Chamber.
8. <u>Comparison of LS-200 and Vonar as a Fire Blocking Layer</u>	In this picture, LS-200 is shown more effective as a fire blocking layer than Vonar when tested with the same thickness.

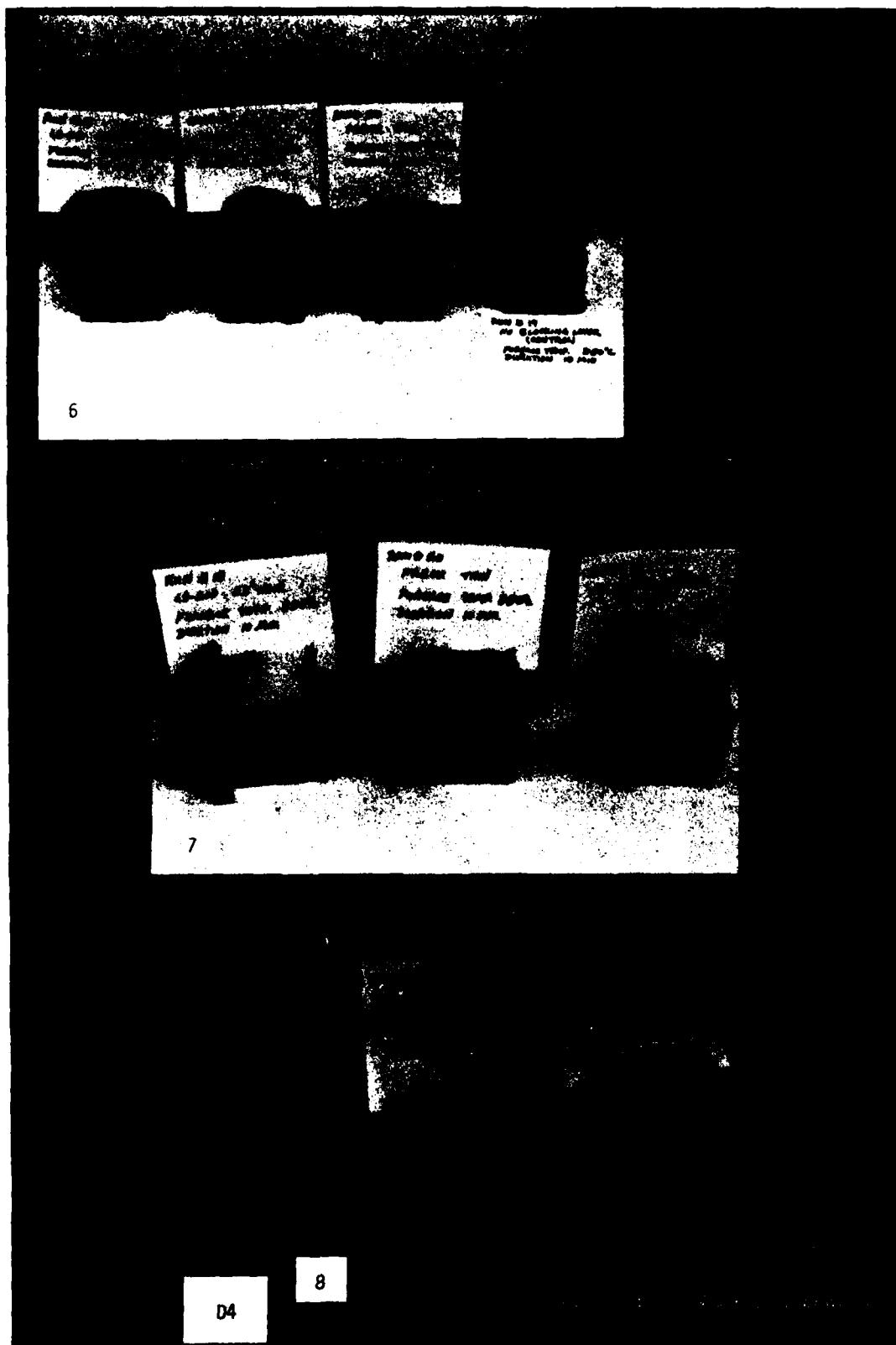


4



5

D3



## APPENDIX E Comparison With Factory Mutual's Numerical Model

The thermochemical model (TCM) of this work is compared with a numerical model developed by Factory Mutual Research Corporation. The numerical approach is described in Appendix A, "A Numerical Model for One-Dimensional Heat Conduction with Pyrolysis in a Slab of Finite Thickness," by F. Tamanini, in Technical Report, FMRC Serial No. 21011.7 dated November, 1976, entitled "The Third Full-Scale Bedroom Fire Test of the Home Fire Project," Vol. II, edited by A.T. Modak.

The computer calculations were actually performed with essentially the same program as listed in the Report, in order to enhance the familiarity with this numerical model (NM). The boundary value read-in instructions were rewritten to suit this comparison purpose, as suggested by the author.

The numerical model (NM) presented in the Report is for a single-layered material: however, the extension to a multi-layered system retaining its features seems possible without difficulty. The change of total weight due to pyrolysis is not included in the original program, but also can be added with ease. Since such extended/modified computer program is unavailable, the comparison of the result of the numerical model with the experimental data of this work is not attempted.

A) Model

$$TCM : k \frac{d^2T}{dx^2} + \rho c_r \frac{dT}{dx} = D \rho N B \exp(-E/RT)$$

$$\frac{dN}{dt} = -NB \exp(-E/RT)$$

$$NM : \frac{\partial}{\partial x} (k_s \frac{\partial T}{\partial x}) + \frac{\partial}{\partial x} (M_g h_g) = Q \frac{\partial \rho_s}{\partial t} + \frac{\partial \rho_s h_s}{\partial t}$$

$$\frac{\partial \rho_s}{\partial t} = -\rho_a B \exp(-E/RT)$$

For NM,  $M_g$  is the mass flux of volatiles,  $h$  is the enthalpy, while subscripts s and g refer to the solid matrix and pyrolysis gas respectively. Another subscript a refers to the unpyrolyzed active material.

TCM	NM
Condensed phase reaction	Same
One - dimensional	Same
Steady state	Unsteady state
Moving coordinate system	Stationary coordinate system
First order Arrhenius reaction for thermal degradation	Same
Analytical solution available	Numerical solution technique presented

### B) Dimensional Analysis

The dimensional analyses are performed for these two models to determine their relative strengths and shortcomings. The dimensionless groups needed to describe these models, according to Pi theorem, could be the following:

Dimensionless Group	JPL's TCM	FMC's NM
Temperature	$\frac{T - T_0}{T_s - T_0}$	$\frac{T - T_0}{T_s - T_0}$
Distance Coordinate	$\frac{\rho_{cx}}{k}$	$\frac{M_{gcx}}{k}$
Heat of Pyrolysis	$\frac{D}{c(T_s - T_0)}$	$\frac{Q}{c(T_s - T_0)}$
Arrhenius Group (or Activation Temperature)	$\frac{E}{RT}$	$\frac{E}{RT}$
Preexponential Factor	$\frac{Bx}{r}$	Bt
Fourier Number	-	$\frac{kt}{\rho_{cx}^2}$

The obvious difference is that Fourier number, which compares a characteristic length dimension with an approximate temperature-wave penetration depth for a given time  $t$ , is included in the NM while it is absent in TCM. Other than the capability of the NM to study unsteady-state behavior, essentially the same dimensionless groups are required for the analysis of the two models.

C) Input Data

	TCM	NM
$C_p, k$	<ol style="list-style-type: none"> <li>1. For one solid</li> <li>2. Stepwise variation with temperature</li> </ol>	<ol style="list-style-type: none"> <li>1. For two solids (active material and char) and one <math>C_p</math> for gas</li> <li>2. Linear variation with temperature</li> </ol>
$\rho$	Stepwise variation with temperature	The initial and final densities (char fraction increases linearly from 0 to 100% of apparent density)
$B, E$	Change often (as evidenced by TGA) depending on temperature range	One set of constant values for the whole temperature range
Heat sink term	D: heat of degradation stepwise variation with temperature (to be obtained from DSC)	Q: heat of pyrolysis fixed at a reference temperature
Boundary Conditions	<ol style="list-style-type: none"> <li>1. Temperature B.C.'s at both boundaries or</li> <li>2. Temperature and heat flux B.C.'s at one boundary</li> </ol>	Either temperature or heat flux B.C. at each of two boundaries

D) Calculation & Results

	TCM	NM
Calculation	Segment-wise application of analytical solution, progressively advancing from one boundary to the other	Finite difference representations of the partial differential equation for the entire region and solving the resultant tridiagonal matrix
Results	surface regression rate, temperature profile and total mass loss	temperature profile, density profile, and mass flux profile

E) Advantages & Disadvantages

	TCM	NM
Advantage	Parametric analysis is readily accessible - essential for basics understanding and establishing design criteria	Time-dependent pyrolysis behavior can be predicted (This may be useful for assuming and confirming time-dependent boundary conditions more realistically.)
Disadvantage	Steady state solution is not adequate for study of transient period	Parametric analysis may be not as clear as TCM

